

ESABASE2 - Debris

Software User Manual

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III. Distribution List

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V. Glossary

Term	Description
Ballistic limit	The minimum particle diameter which is able to penetrate a given wall configuration.
Eclipse	Eclipse is an open source community whose projects are focused on providing an extensible development platform and application frameworks for building software. For detailed information refer to http://www.eclipse.org .
ESABASE	Unix-based analysis software for various space applications. For details refer to the ESABASE User Manual /4/.
ESABASE/Debris	ESABASE framework and the debris and meteoroid flux and damage analysis application.
ESABASE2	New ESABASE version running on PC-based Windows platforms (to be distinguished from the "old" Unix-based ESABASE).
Geometric(al) (analysis)	Flux and damage analysis of a full three-dimensional geometric model.
Georelay	Object pointing keyword: tracking of a GEO satellite.
Ground test	Evaluation of the results of a selected damage or failure equation.
LunarMEM	MEM version, which is tailored to orbits around the Moon.
MASTER 2001	ESA's Meteoroid and Space Debris Terrestrial Environment Reference Model. For details refer to the MASTER Upgrade Final Report /5/. ESABASE2/Debris uses the MASTER 2001 Standard application.
MASTER 2005	ESA's successor to MASTER 2001; now defined as standard application for space debris risk analyses.
MASTER 2009	Successor of ESA's reference model MASTER 2005. For details refer to /24/.
MEM	Meteoroid Engineering Model, for details refer to /20/.
NASA90	Simple analytical space debris engineering model established by NASA /16/.
NASA96 / ORDEM96	NASA's space debris engineering model. Successor of NASA90 and predecessor of ORDEM2000. For details refer to the ORDEM96 documentation /17/.
non-geometric(al) (analysis)	Flux and damage analysis of a plate, which can be specified as a randomly tumbling plate or an oriented plate.
ORDEM2000	NASA's latest space debris engineering model. For details refer to the ORDEM2000 documentation /15/.
STEP	Acronym which stands for the Standard for the Exchange of Product model data.

VI. List of Abbreviations

Abbreviation	Description
ASCII	American Standard Code for Information Interchange
CCSDS	Consultative Committee for Space Data Systems
ECI	Earth Centred Inertial
GUI	Graphical User Interface
IAU	International Astronomical Union
JVM	Java Virtual Machine
LCI	Lunar Centred Inertial
MASTER	Meteoroid and Space debris Terrestrial Environment Reference (Model)
MJD	Modified Julian Day
MLI	Multi-layer insulation
NASA	National Astronautics and Space Administration
OCAF	Open CASCADE Application Framework (contains the ESABASE2 data model)
ORDEM	Orbital Debris Engineering Model
RTP	Randomly Tumbling Plate
UTC	Coordinated Universal Time

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1 Introduction

ESABASE2 is a software application (and framework) for space environment analyses, which play a vital role in spacecraft mission planning. Currently (2012), it encompasses Debris/meteoroid /1/, Atmosphere/ionosphere /7/, Contamination/outgassing /5/, /6/ and Sunlight /8/ analyses; with this, it complements other aspects of mission planning like thermal or power generator design.

The application grew from ESABASE2/Debris, an application for space debris and micro-meteoroid impact and damage analysis, which in turn is based on the original ESABASE/Debris software /4/ developed by different companies under ESA contract. ESABASE2 adds a modern graphical user interface enabling the user to interactively establish and manipulate three-dimensional spacecraft models and to display the selected orbit. Analysis results can be displayed by means of the colour-coded surfaces of the 3D spacecraft model, and by means of various diagrams.

The development of ESABASE2 was undertaken by etamax space GmbH under the European Space Agency contract No. 16852/02/NL/JA. The first goal was to port ESABASE/Debris and its framework/user interface to the PC platform (Microsoft Windows) and to create a modern user interface.

From the start, the software architecture has been expressively designed to accommodate further applications: the solvers outlined in the first paragraph were added, and more modules like e.g. Radiation are to follow.

ESABASE2 is written in Fortran 77, ANSI C++ and Java 6. The GUI is built on top of the Eclipse rich client platform, with 3D visualisation and STEP import realised by Open CASCADE. Report and graphs are based on the JFreeReport/JFreeChart libraries.

This user manual is the **Debris handbook**. It complements the Framework user manual /1/, which explains the common functionality of all solvers (e.g. Debris, Sunlight, Atmosphere/Ionosphere, or COMOVA).

2 Debris Solver

After we have specified mission and spacecraft geometry /1/, the next step is to perform space environment analyses with these. One of the available solvers, ESABASE2/Debris, performs debris and meteoroid analyses within the framework.

Five space debris models (NASA90 /16/, ORDEM2000 /15/, MASTER 2001 /11/, MASTER 2005 /19/ and MASTER 2009 /23/) as well as four meteoroid models (Grün /14/, Divine-Staubach /13/, /18/, and MEM /20/ respective LunarMEM /25/) are currently available for flux and damage analysis.

For detailed information on the technical background of the space debris and micro-meteoroid simulation, please refer to the ESABASE2/Debris technical description /2/.

This debris solver chapter is structured as follows:

- *Debris Geometry*: Explains debris-specific additions to a S/C geometry (see section 2.1).
- *Debris Input*: Describes the input parameters for Debris and meteoroid analyses (see section 2.2).
- *Debris Analysis*: How to perform an analysis (see section 2.3).
- *Debris Results*: How to interpret the analysis results (see section 2.4).

2.1 Debris Geometry

The geometry editor defines a spacecraft's geometry for all solvers available within the ESABASE2 framework /1/, including Debris. Solver-specific geometry parameters are defined using special pages in the shape wizard.

For Debris, the following special pages and options are available:

- *Debris Page*: Settings only applicable to the Debris solver, concerning material thickness and shielding (see section 2.1.1).
- *Meshing Page*: Positive/negative sides and meshing recommendations for Debris (see section 2.1.2).

2.1.1 Debris Page

A dedicated Debris page is shown for each shape, allowing you to define the shielding configuration of any shape (both primary and secondary shielding). Classic surface material properties as defined by the Material page are not interpreted by the Debris solver.

Please note that the Debris page is only available if ESABASE2/Debris is part of your installation (it is also possible to have only ESABASE2/Atmosphere, for example, depending on your license).

To see the Debris page, open a geometry file, select a shape, then rightclick it and choose "Modify → Debris". A wizard page as shown in the figure below will be opened.

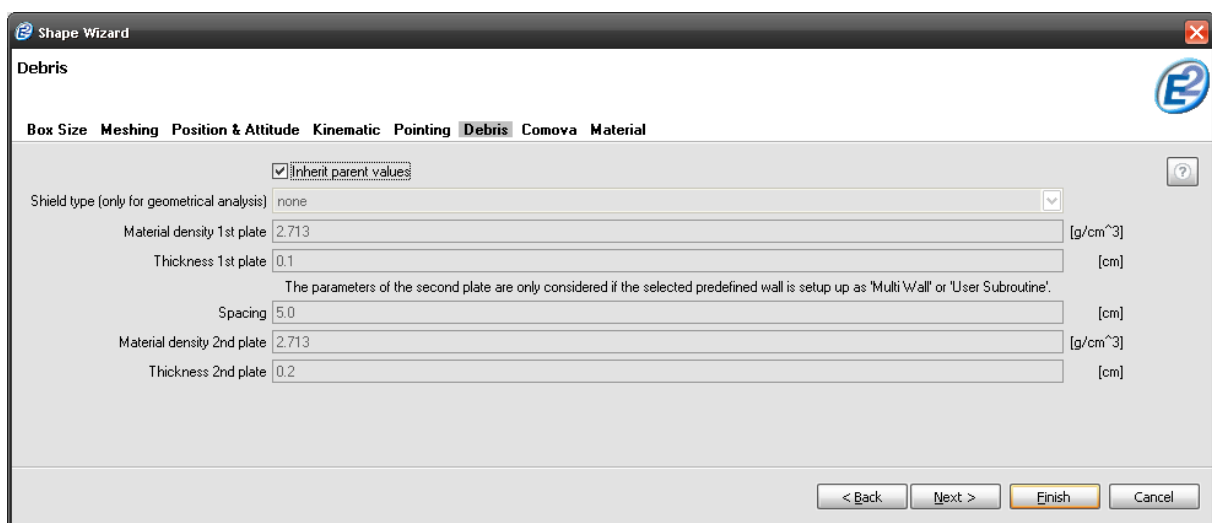


Figure 2.1: Geometry editor, Debris page

On this page, you can define shielding and material parameters that go beyond the Material page of the shape wizard.

The simplest option is to check the "Inherit parent values" checkbox, which takes over all Debris related values from the parent shape. This is possible for all shapes except the central body.

Below, the shield type can be chosen. One of the 15 predefined damage and failure equation combination can be selected here (Wall_1 to Wall_15, for setup see 2.2.1.4). The lower three parameters (spacing, Material density 2nd plate and Thickness 2nd plate) are only used if the selected wall is defined as multiwall.

In the screenshot above, you see "none" chosen and thus all further parameters are disabled.

The figure below illustrates the parameters from the Wizard.

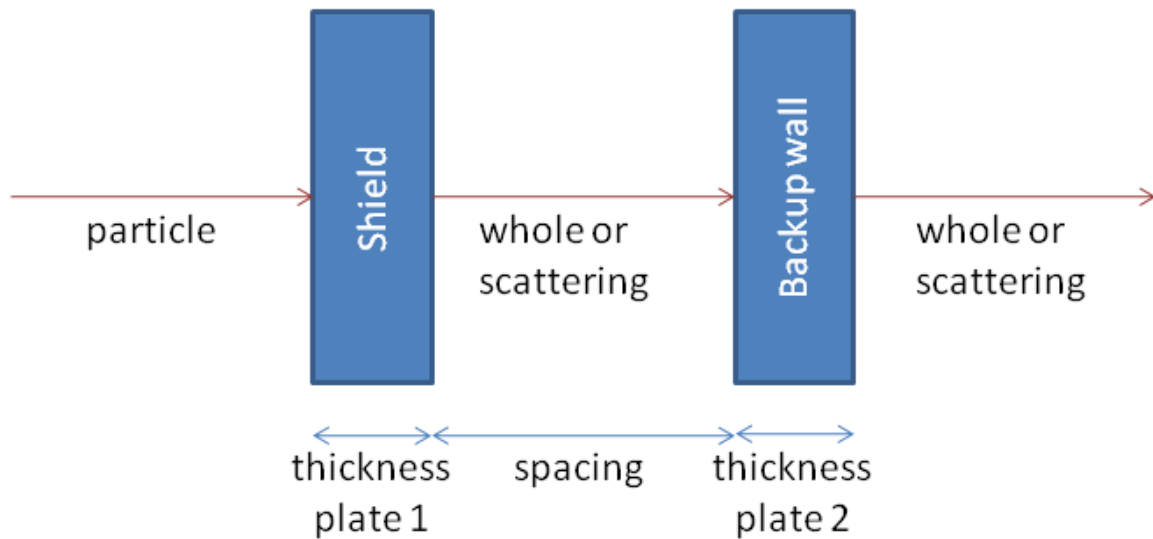


Figure 2.2: Debris shielding parameters

A particle hits the first shield wall, and the effect depends – besides the impact velocity, the impact angle and particle properties such as its diameter and material density – on the material density and thickness of the wall. The particle is either stopped, or penetrates the wall; it can remain whole or be scattered into smaller pieces due to the impact. If it penetrates, it travels the spacing between the walls; the same stop/penetration/scatter happens with the second wall. Depending on the remaining energy, the particle causes (a) a crater on or (b) a penetration of the device behind the wall.

In a single-wall scenario, the second wall does not exist, and the likelihood of a crater or penetration is considerably increased.

A failure equation determines whether a particle penetrates the wall configuration. A damage equation determines the size of the crater or hole (depending on no penetration/penetration) on the first wall (shield).

More information can be found in the ESABASE2/Debris technical description /2/ and in the IADC Protection Manual /21/.

2.1.2 Meshing Page

Although the meshing page (shown below) is the same for all solvers, the effects of the settings may vary.

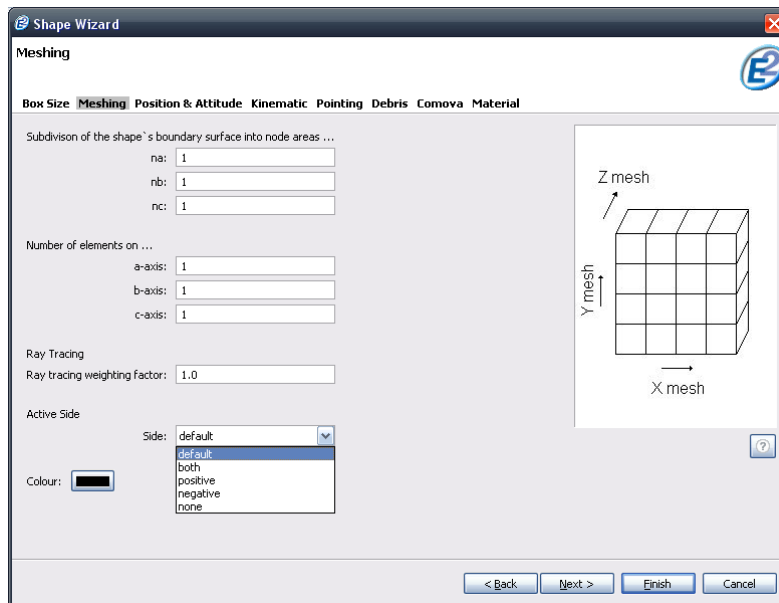


Figure 2.3: Geometry editor, Meshing page

Subdivisions of surfaces (i.e. "surface nodes") are not supported by Debris. We recommend using the "Number of elements on" settings for finer meshing.

Regarding active sides: Debris results always comprise both sides of an element (the solver sums them up). For volumes, this is uncritical, because only one side is ever affected. For surfacic shapes (e.g. plate), we recommend choosing only one active side, because the results can otherwise not be correctly assigned to an element side.

Please note: In NASA90, the average (instead of the sum) of both sides is taken. This is a known bug.

And with this, the geometry file is ready, and we can move to the Debris input editor.

2.2 Debris Input

Complementing the Debris parameters bound to the geometry (see previous section), the global Debris/Meteoroid input parameters are all specified in the Debris input editor. This editor is divided into three tabs:

- *Debris Main Tab*: Specifies the parameters for the geometrical analysis (see section 2.2.1).
- *Ground Test Tab*: An efficient way to test damage equations (see section 2.2.2).
- *Non-Geometric Analysis Tab*: Allows a fast guess for the expected flux values on a specific orbit (see section 2.2.3).

2.2.1 Debris Main Tab

The main tab of the Debris input editor contains four major blocks for specifying geometrical analysis input parameters:

- *Model selection*: Allows you to choose among debris and meteoroid models, and to edit dedicated model parameters (see section 2.2.1.1).
- *Size boundaries*: Limits on the type of debris or meteoroids to be considered in the analysis (see section 2.2.1.2).
- *Ray tracing*: Defines the accuracy of ray tracing results (see section 2.2.1.3).
- *Damage Model*: Defines a set of failure and damage equation combinations for the Debris analysis (see section 2.2.1.4).

Within the damage model block the option "user subroutine" can be selected. How to define your own damage equation in a Fortran library will be explained in section 2.2.1.5.

The figure below shows the ESABASE2/Debris main tab within the Debris input editor.

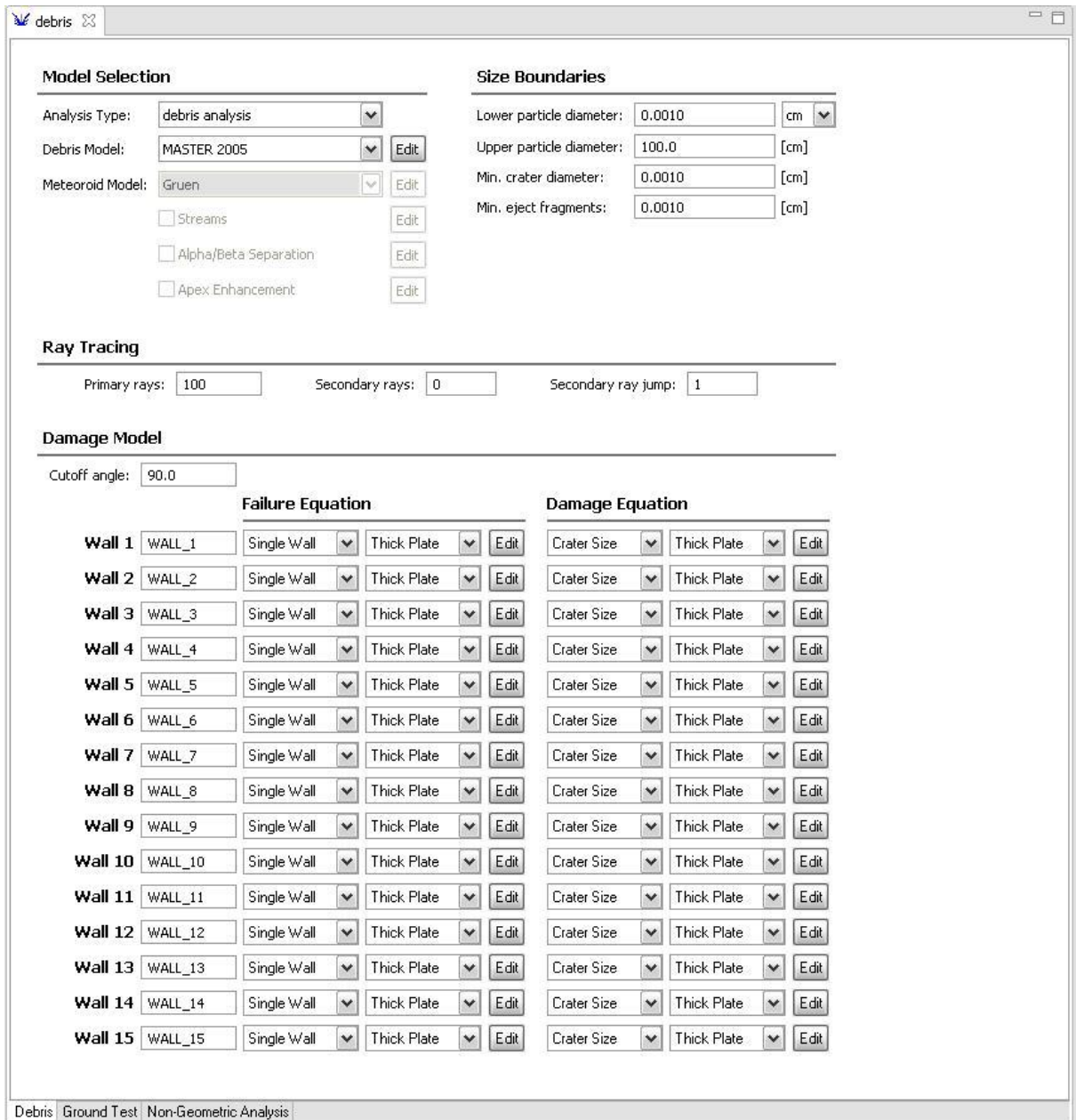


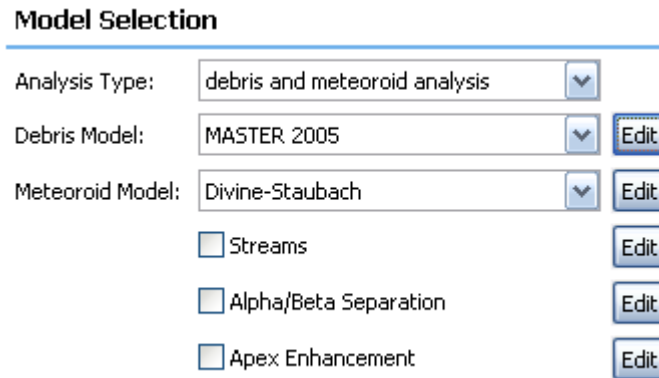
Figure 2.4: Debris input editor, main tab

At the bottom of the editor, you can see the "Debris", "Ground Test" and "Non-Geometric Analysis" tabs. This subsection is concerned with the main "Debris" tab.

In the following, the four sections of the main tab will be explained.

2.2.1.1 Model selection

Your first decision is which debris and meteoroid models you want to use. The following figure shows the model selection block within the Debris main tab.



Model Selection

Analysis Type: debris and meteoroid analysis

Debris Model: MASTER 2005 Edit

Meteoroid Model: Divine-Staubach Edit

Streams Edit

Alpha/Beta Separation Edit

Apex Enhancement Edit

Figure 2.5: Debris input editor, main tab, Model Selection

As a starting point to the Debris analysis, you have the choice between debris or meteoroid analyses, or both (see first combo-box: "Analysis Type"). Depending on your choice, the "Debris Model" or "Meteoroid Model" combo boxes will be enabled or disabled.

ESABASE2/Debris provides five Debris models (MASTER 2001, MASTER 2005, MASTER 2009, ORDEM 2000, NASA90) and four meteoroid models (Grün, Divine-Staubach, MEM, LunarMEM). Most of the models accept detailed input parameters: Press the "Edit" button to the right of the model combo-boxes.

Additional parameters (Streams, Alpha/Beta Separation, and Apex Enhancement) are shown below; they are valid only for some models. Details are given in the sections 2.2.1.1.10, 2.2.1.1.11 and 2.2.1.1.12.

For the analysis on lunar orbits only the two meteoroid models Grün and LunarMEM are applicable (no debris models will be accepted). In combination with a trajectory file the debris model MASTER 2009 and the meteoroid models Grün, MEM and LunarMEM are applicable. The use of other models for lunar orbits or with a trajectory files than listed here, causes a stop of the analysis execution informing the user about the invalid configuration.

In the following, the available models will be described together with their associated parameters. This description is necessarily short; a full description can be found in the ESABASE2/Debris Technical Description /2/.

2.2.1.1.1 Debris model: MASTER 2001

MASTER 2001 /11/ is the 2001 version of ESA's meteoroid and space debris reference model; it is the forerunner of MASTER 2005 (see next subsection). When you click on the "Edit" button, a dialog with MASTER 2001 input parameters will open, as shown in the following figure.

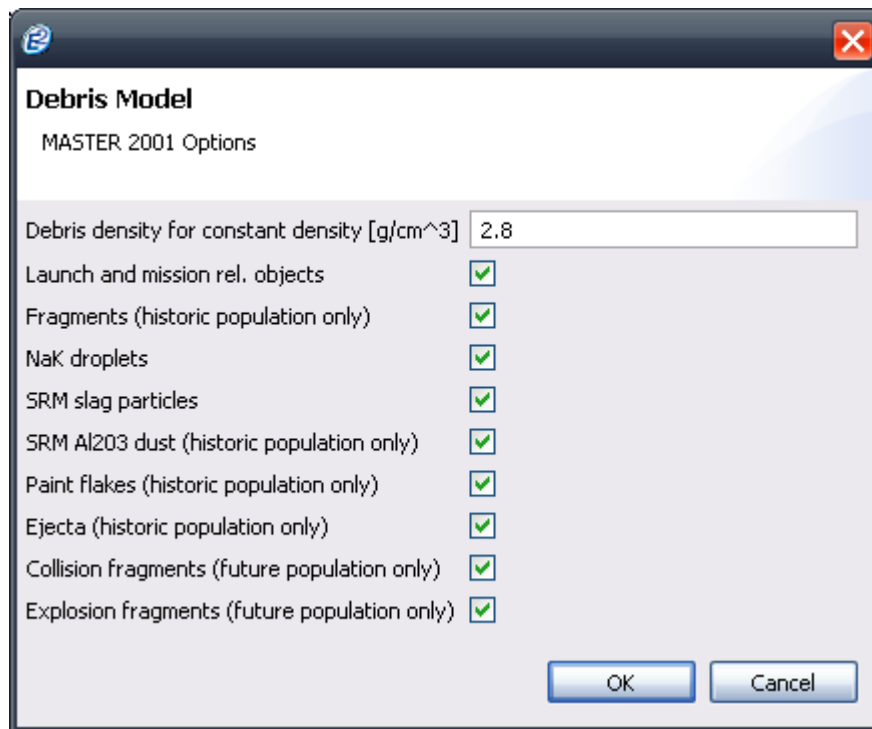


Figure 2.6: Debris input editor, main tab, Model Selection, MASTER 2001

Apart from the assumed debris density (default: 2.8 g/cm³ as material mix average), the MASTER 2001 model is based on numerical modelling of various population sources, which can be included or excluded from an analysis.

- Launch and mission related objects: payloads and satellites, upper stages, support structures. These are mostly larger, trackable objects.
 - Note that the Westford needles experiment is included as subpopulation and cannot be turned off with this flag.
- Fragments (collision and explosion) before reference epoch (2001-05-01): These are known fragment populations.
- NaK droplet releases: coolant droplets released from Russian RORSAT satellites.
- SRM (solid rocket motor) firing waste products:
 - slag produced in the final firing phase, mostly > 1 mm

- Aluminium oxide (Al₂O₃) dust
- Paint flakes are generated by surface degradation effects (mostly sunlight and thermal cycling)
- Ejecta are small fragments of the S/C created by the impact of debris or meteoroids.
- Collision (but not explosion) fragments after reference epoch; covers assumed collision rate of satellites with other bodies in the future.
- Explosion (but not collision) after reference epoch; covers assumed explosion rate of satellites in the future.

In the context of the MASTER 2001 model, "reference epoch" or "historic" means dates until 2001-05-01. The "future" are dates from 2001-05-01; note that from there, objects < 1 mm are not considered (this also means that paint flakes, ejecta and dust are not available, because they are always < 1 mm).

As an example of the effect of the various factors, the figure below shows the cumulative cross-sectional debris flux from different sources on an ISS-like orbit.

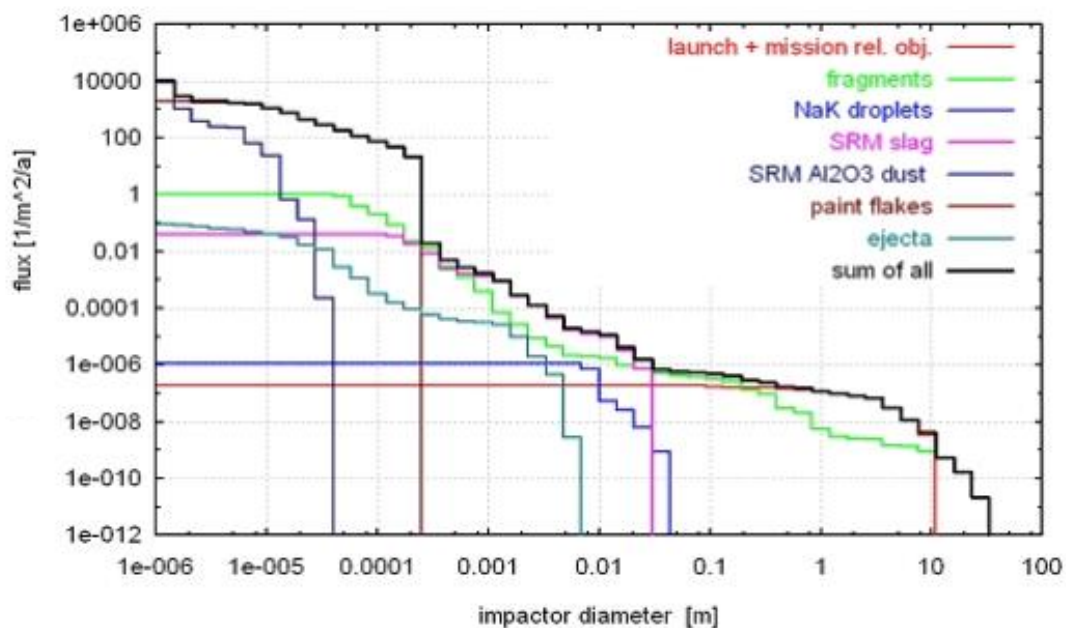


Figure 2.7: Cumulative flux vs. particle diameter for an ISS-like orbit (MASTER 2001)

MASTER 2001 provides realistic yearly population snapshots for the past and the future. The flux calculation is based on the analytic evaluation of the distributions of the size and the orbital elements of the particle population (MASTER 2001 Standard application).

The model considers the population asymmetry induced by the asymmetric distribution of the particle orbits argument of perigee.

MASTER 2001 covers the entire altitude range from LEO to GEO (150 km to 37000 km). Within the given altitude range there are no restrictions concerning the target orbit, so that highly eccentric orbits such as GTO can be analysed.

The MASTER population snapshots are available from year 1980 to 2020. Consequently, historic missions (e.g. LDEF) and future missions can be analysed using realistic population snapshots. The population snapshots of the May 1st of the mission start year are used for the analysis.

For the future evolution of the space debris environment the following assumptions have been applied:

- continuation of space activity (launches, explosions, solid rocket motor firings) at the same rate as in the recent past,
- no new satellite constellations deployed,
- no implementation of debris mitigation measures.

This corresponds to the MASTER 2001 future reference scenario.

2.2.1.1.2 Debris model: MASTER 2005

MASTER 2005 /19/ is ESA's meteoroid and space debris reference model, and the successor of MASTER 2001 (see previous subsection). When you click on the "Edit" button, a dialog with MASTER 2005 input parameters will open, as show in the following figure.

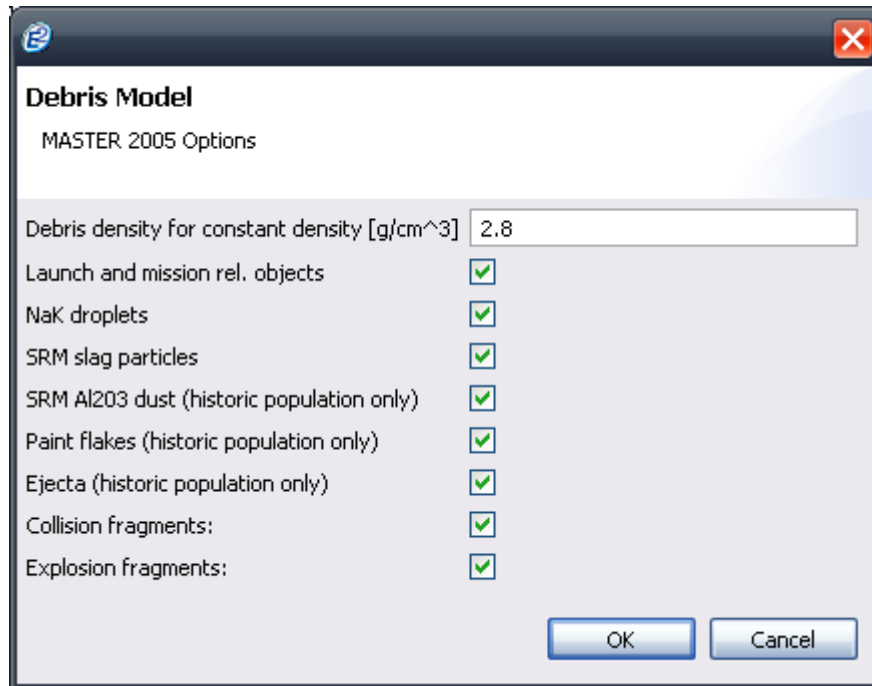


Figure 2.8: Debris input editor, main tab, Model Selection, MASTER 2005

The MASTER 2005 options are similar to the MASTER 2001 options (see previous subsection). The following differences have to be noted:

- Collision and explosion fragments cover both historic and future populations, not only historic populations as in MASTER 2001.
 - This also explains why the "Fragments (historic)" option has vanished in comparison to MASTER 2001.

Like MASTER 2001, MASTER 2005 covers altitudes from 186 km to 37000 km. The "future" is from 2005-05-01 and in the future; objects < 1 mm are not considered. The population snapshots of the May 1st of the mission start year are used for the analysis.

2.2.1.1.3 Debris model: MASTER 2009

MASTER 2009 /23/ is ESA's meteoroid and space debris reference model, and the successor of MASTER 2005 (see previous subsection). When you click on the "Edit" button, a dialog with MASTER 2009 input parameters will open, as show in the following figure.

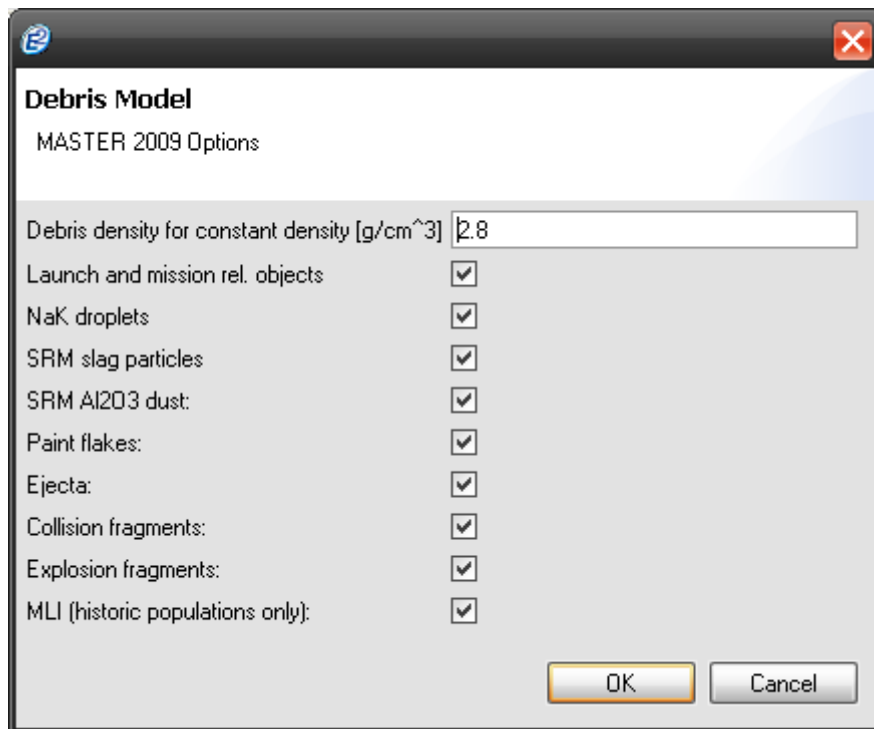


Figure 2.9: Debris input editor, main tab, Model Selection, MASTER 2009


The MASTER 2009 options are similar to the MASTER 2005 options (see previous subsection). The following differences to MASTER 2005 have to be noted:

- Objects down to 1 μm are considered for both the historic and the future populations (MASTER 2005 do not consider objects < 1 mm for future populations)
- Both historic and future populations for solid rocket motor dust are available \rightarrow absence of the hint 'historic populations only'.
- Both historic and future populations for paint flakes are available \rightarrow absence of the hint 'historic populations only'.
- Both historic and future populations for ejecta are available \rightarrow absence of the hint 'historic populations only'.
- The new source multi layer insulation (MLI) is introduced. Historic populations until the reference date are provided for this source.

The reference date of MASTER 2009 is the 2009-05-01. Dates after the reference date are considered as "future". MASTER 2009 covers altitudes from 186 km to ca. 37000 km. The population snapshots of the May 1st of the mission start year are used for the analysis.

2.2.1.1.4 Debris model: NASA90

NASA90 /16/ is an analytical debris model developed by NASA, which provides a simple and very fast debris flux calculation, but does not fully reflect the current knowledge of the Earth's debris environment, in particular the existence of a large number of particles on eccentric orbits.

Upon clicking the "Edit" () button, the NASA90 input parameters dialog is opened, as shown in the screenshot below.

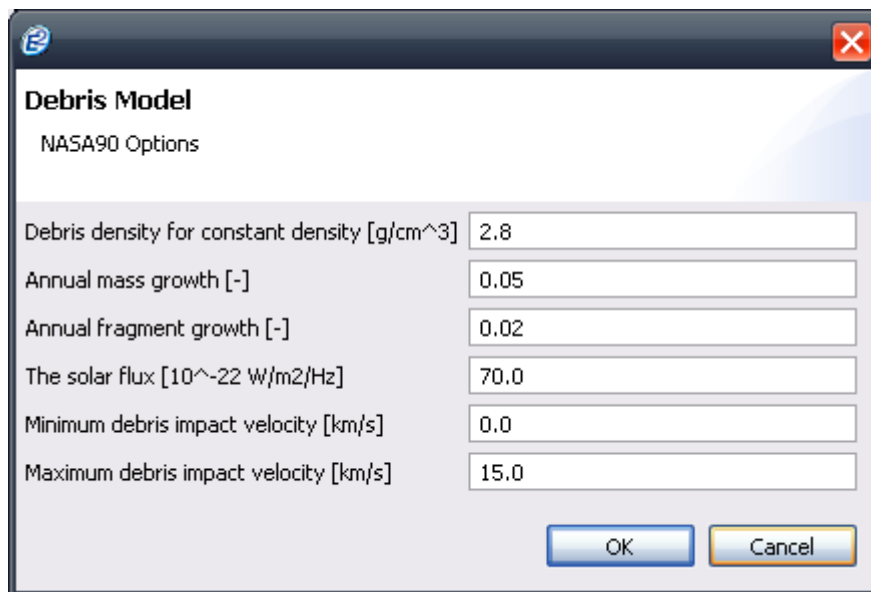


Figure 2.10: Debris input editor, main tab, Model Selection, NASA90

As with all debris models, the average density of the debris material can be specified.

The debris mass (P) and fragments number (Q) annual growth rates are specified as percentage, where 1 = 100% growth. It is recommended to use the default values.

Solar flux is used to specify the solar activity; appropriate values for the mission duration can e.g. be retrieved using the publicly available SWENET database /22/.

At the bottom, you see the debris velocity range, namely the minimum and maximum debris impact velocity to be considered. The default is from 0 to 20 km/s.

Please note that the NASA90 model is restricted to orbital altitudes below 1000km.

2.2.1.1.5 Debris model: ORDEM 2000

ORDEM2000 /15/ is a debris model developed by NASA; it is the successor of NASA96, which in turn followed NASA90. The model describes the orbital debris environment in the low earth orbit region between 200 km and 2000 km altitude.

The only user-editable input parameter is the assumed debris material density (default: 2.8 g/cm³), as shown in the figure below.

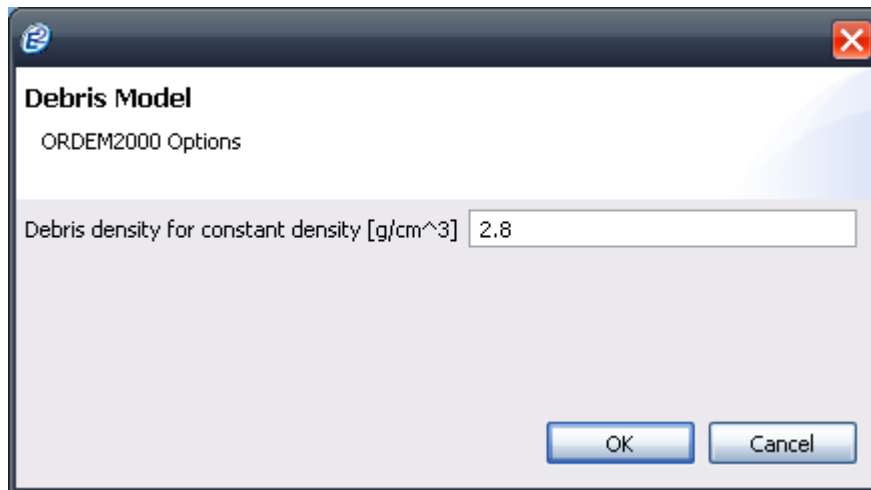


Figure 2.11: Debris input editor, main tab, model selection, ORDEM 2000

ORDEM 2000 is appropriate for engineering solutions requiring knowledge and estimates of the orbital debris environment (debris spatial density, flux, etc.). The model includes a large set of observational data (both in-situ and ground-based), covering the object size range from 10 µm to 10 m.

The analytical technique uses a maximum likelihood estimator to convert observations into debris population probability distribution functions; these functions then form the basis of the debris populations. A finite element model processes the debris populations to form the debris environment.

2.2.1.1.6 Meteoroid model: Grün

Grün /14/ is an omni-directional interplanetary flux model for sporadic meteoroid environment. It can be applied for Earth and lunar orbits. When you press the "Edit" button, a dialog with the Grün input parameters shown in the figure below appears.

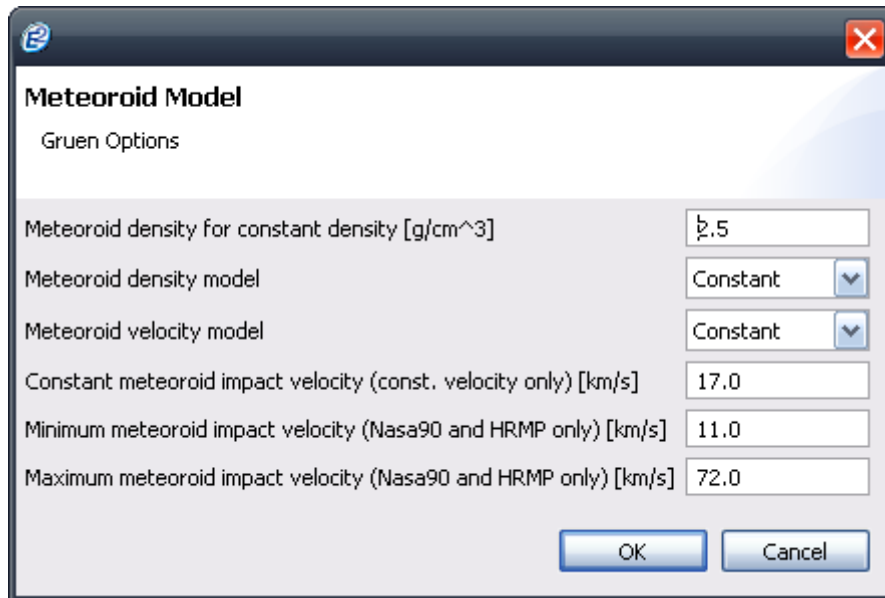


Figure 2.12: Debris input editor, main tab, Model Selection, Grün

On top, you see the meteoroid density option. Below, choose between this constant meteoroid density and alternatively the NASA90 density distribution model (the latter choice will ignore your meteoroid density specification above).

The next two lines are concerned with the meteoroid velocity distribution option:

- Constant meteoroid velocity; for this option, the default value is 17 km/s.
- The NASA90 velocity distribution. In most cases, this delivers the best results, and is thus the recommended option for the industrial user. The NASA90 velocity distribution is not applicable to lunar orbits, due to the intended design for Earth orbits.
- The Taylor HRMP velocity distribution. This model is the most complex option. It is also applicable and recommended for the lunar orbits.

The last two lines handle the meteoroid velocity range, namely the minimum and maximum meteoroid velocity to be considered. The default is 11 km/s and 72 km/s.

2.2.1.1.7 Meteoroid model: Divine-Staubach

Divine-Staubach /13/ /18/ is a meteoroid model which is also part of the MASTER 2005 model. When you press the "Edit" button, a dialog shows the the Divine-Staubach input parameters depicted in the screenshot below.

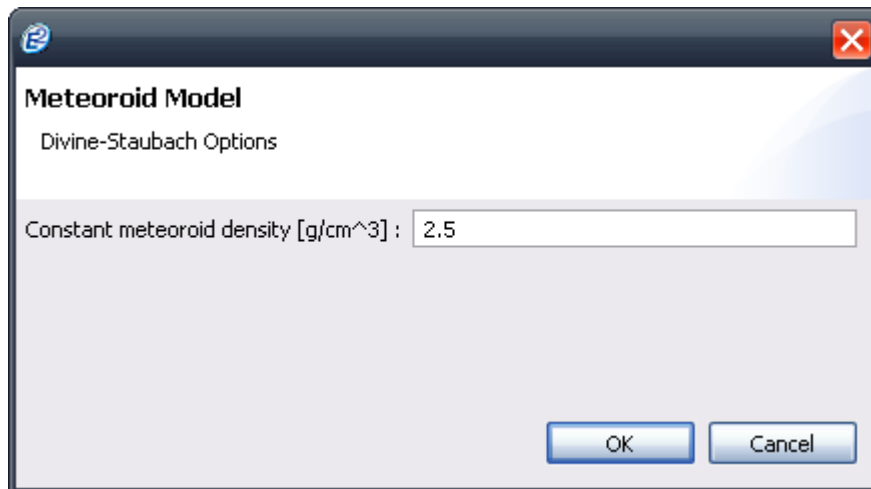


Figure 2.13: Debris input editor, main tab, Model Selection, Divine-Staubach

The only parameter is the material density of meteoroids. It is assumed to be constant.

Divine-Staubach is based on the size and orbital element distributions of five meteoroid sub-populations, and thus provides directional information in the same way as the MASTER 2005 debris model.

Please note: Since the Divine-Staubach meteoroid model is implemented in the MASTER Standard application, one MASTER 2005 run covers both debris and meteoroid flux determination if the corresponding switches are set.

2.2.1.1.8 Meteoroid model: MEM

MEM /20/ is a meteoroid model developed by the University of Western Ontario, Canada. Upon pressing "Edit", a dialog with the MEM input parameters opens, as shown in the figure below.

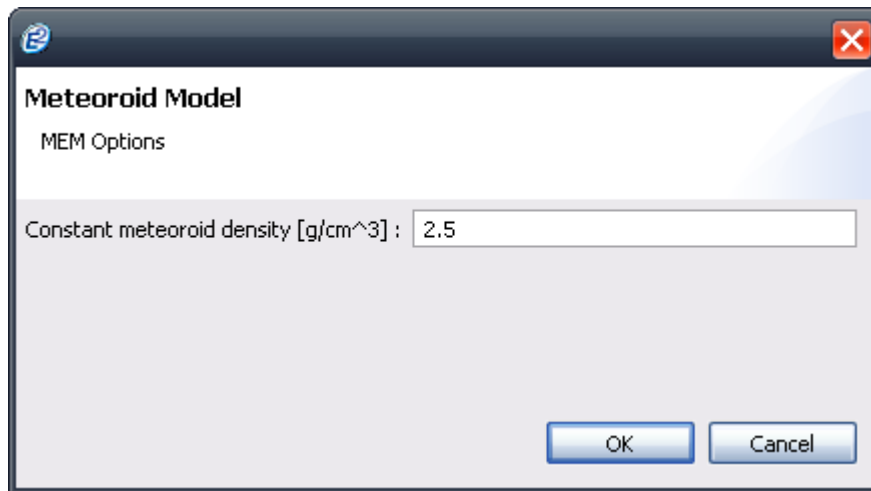


Figure 2.14: Debris input editor, main tab, Model Selection, MEM

MEM has no options by itself, using a constant meteoroid density of 1 g/cm^3 . The option available here is used to convert mass to diameter for damage computation. We propose to always use 1 g/cm^3 .

Please note that the default 2.5 g/cm^3 is fitting only for the other meteoroid models – all models use the same parameter within the data model; this is the reason that the default in the GUI cannot be 1 g/cm^3 .

MEM is a parametric model of the spatial distribution of sporadic meteoroids. The primary source is short-period comets with aphelia less than 7 AU. The model also considers the contributions from long-period comets to the sporadic meteor complex, and includes the effects of the gravitational shielding and focussing of the planets.

2.2.1.1.9 Meteoroid model: LunarMEM

LunarMEM is a MEM /20/ version, which is tailored to the vicinity of the Moon. Upon pressing "Edit", a dialog with the LunarMEM input parameters opens, as shown in the figure below.

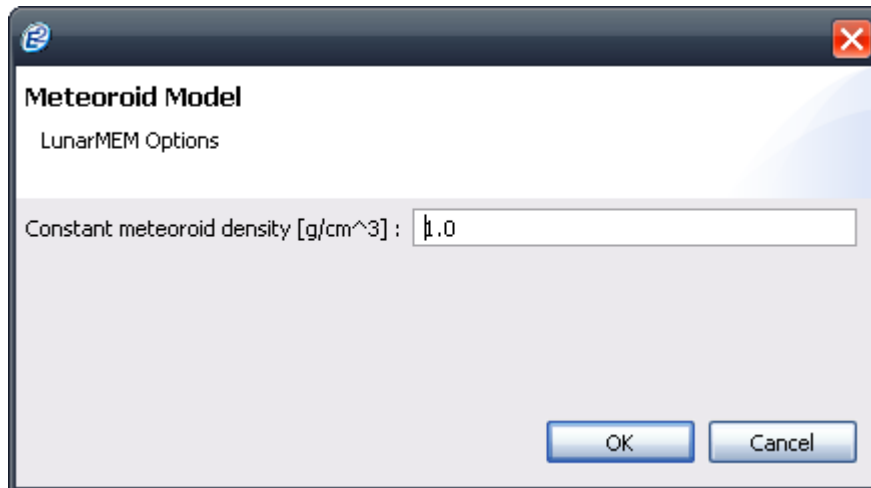


Figure 2.15: Debris input editor, main tab, Model Selection, LunarMEM

Since LunarMEM is a tailored version of MEM most model characteristics are coherent, as for the use of a constant meteoroid density of 1 g/cm^3 . We propose to always use 1 g/cm^3 also for LunarMEM.

Please note that the default 2.5 g/cm^3 is fitting only for the other meteoroid models – all models use the same parameter within the data model; this is the reason that the default in the GUI cannot be 1 g/cm^3 .

For further information about LunarMEM please refer to /2/ or the MEM reference /20/.

2.2.1.1.10 Jenniskens Stream Model

The above meteoroid models are describing the background meteoroid flux; the Jenniskens stream model adds annual meteoroid streams (e.g. the Perseids). The streams model can be used with or without another meteoroid model, only for the analyses on Earth orbits.

When pressing the "Edit" button, the dialog for the Jenniskens Stream Model input parameters appears, as shown in the figure below.

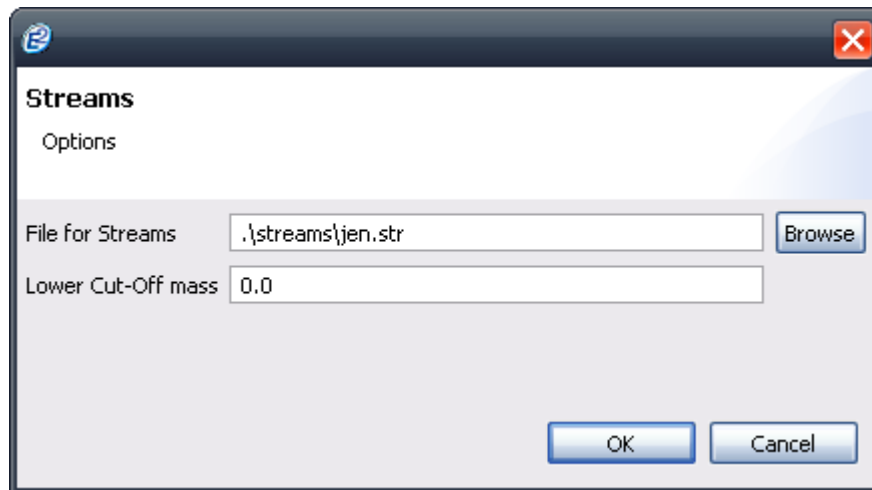


Figure 2.16: Debris input editor, main tab, Model Selection, Streams

The streams and possible interstellar sources are defined in an external input file, with a default `jen.str` being provided by ESABASE2. File locations are always relative to the ESABASE2 installation path.

Meteoroid particles with less mass than the lower cut-off mass are not considered in the analysis.

The Jenniskens stream model is based on observation data gathered over a 10 year period, and can be applied to flux and damage analysis. It also includes directional information on the streams.

Please note that for long mission durations, the directional effect is smeared out, and does not give any additional information than the Grün sporadic option. The stream option is best suited for the investigation of missions below 10 days where high stream activities are expected.

2.2.1.1.11 Alpha-Beta Separation

As an improvement to the Grün meteoroid model, the Alpha-Beta Separation divides the meteoroids in alpha particles (following the Grün sporadic omni-directional flux model) and smaller beta particles stemming from the sun.

When you press "Edit", the Alpha-Beta Separation options shown in the screenshot below appear in a dialog.

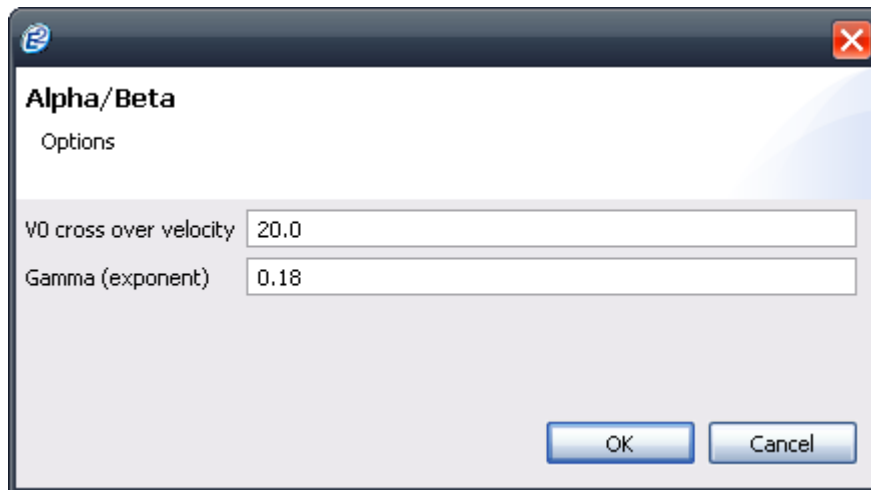


Figure 2.17: Debris input editor, main tab, Model Selection, Alpha-Beta Separation

The velocity distribution of the beta particles can be modified by specifying two parameters (for more details, see /10/):

- V0: cross-over velocity; default: 20.0 km/s
- Gamma (exponent); default: 0.18

The Alpha-Beta Separation obtains improved directional information by attempting to split off the beta meteoroids, which are driven away from the Sun into hyperbolic orbits by radiation pressure, from the alpha meteoroids. It is applicable only to Earth orbits.

An apex enhancement of the alpha meteoroids and interstellar streams (see next subsection) may introduce further directional information.

We do not recommend using other values than the default values for this model, unless you have in-depth knowledge in astrophysics.

2.2.1.1.12 Apex Enhancement

As a modification to the Grün model, the Apex Enhancement describes the meteoroid flux enhancement caused by the earth's motion on its orbit around the sun (similar to the enhanced flux a spacecraft experiences in velocity direction). It is applicable only to Earth orbits.

If you press "Edit", a dialog shows the Apex Enhancement input parameters shown in the figure below.

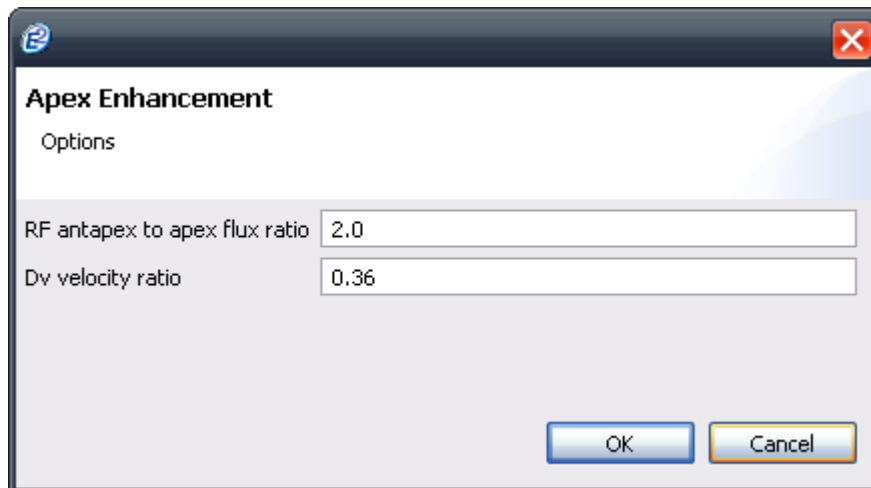


Figure 2.18: Debris input editor, main tab, Model Selection, Apex Enhancement

Two parameters can be used to describe the ratio of flux and velocities between apex ("front") and antapex ("back") directions (see /10/ for more details):

- R_F : antapex to apex flux ratio; default: 2.0
- D_V : velocity ratio; default: 0.36

As with Alpha-Beta Separation, we do not recommend using other values than the default values for this model, unless you have in-depth knowledge in astrophysics.

2.2.1.2 Size Boundaries

The figure below shows the Size Boundaries section of the main tab in the Debris input editor. In this section, you specify limits on the type of Debris or Meteoroids to be used in the analysis.

Size Boundaries



Lower particle diameter:	<input type="text" value="0.0010"/>	 
Upper particle diameter:	<input type="text" value="100.0"/>	[cm]
Min. crater diameter:	<input type="text" value="0.0010"/>	[cm]
Min. eject fragments:	<input type="text" value="0.0010"/>	[cm]

Figure 2.19: Debris input editor, main tab, Size Boundaries

At the top, the minimum and maximum particle size limits can be input as mass [g] or diameter [cm]. The conversion between mass and diameter assumes spherical particles.

At the bottom, the minimum crater diameter and ejecta fragment size are defined in [cm]. Below these size limits, crater or ejecta fragment are not further considered in the analysis.

2.2.1.3 Ray Tracing

Ray tracing is the primary technique in ESABASE2 to determine whether Debris or Meteoroids hit the spacecraft geometry at a specific orbital point. Below, you can see a screenshot of the Ray Tracing section in the main tab.



Figure 2.20: Debris input editor, main tab, Ray Tracing

The "Primary rays" parameter governs the number of primary rays to be fired per element. For ESABASE geometric models, at least 250 rays per element are recommended, for non-geometric analyses 1000 rays.

In the middle, the "Secondary rays" parameter specifies the number of secondary rays to be fired from each impact point. Due to the high computational effort caused by this option it is recommended to choose fairly low values (< 100).

Another option to reduce computation time is to specify a "Secondary ray jump" which causes the program to skip the respective number of secondary rays.

2.2.1.4 Damage Model

Up to 15 different damage models can be defined for damage assessment. Each model is defined by two types of damage equations which are used during the analysis:

- The ballistic limit equations (or failure equation), which deliver the limit impactor diameter above which a structural failure of the analysed surface occurs. This limit diameter is used to compute the probability of a failure using the chosen flux models.
- The damage size equations, which compute the crater diameter of semi-infinite targets and the hole size of thin targets. Again, the flux models are then used to compute the total cratered or perforated area of the surface.

The figure below shows the damage model setup in the main tab.

Damage Model

Cutoff angle:

	Failure Equation			Damage Equation		
Wall 1	<input type="text" value="WALL_1"/>	Single Wall	Thick Plate	Crater Size	Thick Plate	Edit
Wall 2	<input type="text" value="WALL_2"/>	Single Wall	Thick Plate	Crater Size	Thick Plate	Edit
Wall 3	<input type="text" value="WALL_3"/>	Single Wall	Thick Plate	Crater Size	Thick Plate	Edit
Wall 4	<input type="text" value="WALL_4"/>	Single Wall	Thick Plate	Crater Size	Thick Plate	Edit
Wall 5	<input type="text" value="WALL_5"/>	Single Wall	Thick Plate	Crater Size	Thick Plate	Edit
Wall 6	<input type="text" value="WALL_6"/>	Single Wall	Thick Plate	Crater Size	Thick Plate	Edit
Wall 7	<input type="text" value="WALL_7"/>	Single Wall	Thick Plate	Crater Size	Thick Plate	Edit
Wall 8	<input type="text" value="WALL_8"/>	Single Wall	Thick Plate	Crater Size	Thick Plate	Edit
Wall 9	<input type="text" value="WALL_9"/>	Single Wall	Thick Plate	Crater Size	Thick Plate	Edit
Wall 10	<input type="text" value="WALL_10"/>	Single Wall	Thick Plate	Crater Size	Thick Plate	Edit
Wall 11	<input type="text" value="WALL_11"/>	Single Wall	Thick Plate	Crater Size	Thick Plate	Edit
Wall 12	<input type="text" value="WALL_12"/>	Single Wall	Thick Plate	Crater Size	Thick Plate	Edit
Wall 13	<input type="text" value="WALL_13"/>	Single Wall	Thick Plate	Crater Size	Thick Plate	Edit
Wall 14	<input type="text" value="WALL_14"/>	Single Wall	Thick Plate	Crater Size	Thick Plate	Edit
Wall 15	<input type="text" value="WALL_15"/>	Single Wall	Thick Plate	Crater Size	Thick Plate	Edit

Figure 2.21: Debris input editor, main tab, Damage Model

There are no conventions concerning the combination of failure and damage equation. Single wall as well as multi wall can be combined with crater or hole size equations.

Failure and damage equations of the ESABASE2/Debris analysis tool are defined in a parametric form, with editable parameters (constants and exponents). The parameters can be edited via the "Edit" buttons shown in Figure 2.21.

Up to seven entities can be defined:

for failure equation:

- the single wall ballistic limit equation
- the multiple wall ballistic limit equation and
- the user subroutine parameters

for damage equation:

- the crater size equation
- the parametric clear hole equation
- the advanced hole equation and
- the user subroutine parameters.

The user subroutine option is only for expert users with Fortran and/or C++ experience (see section 2.2.1.5).

For a detailed description of the user input of the damage equations, please refer to /10/.

2.2.1.5 User Subroutine

For expert users of the original ESABASE, the possibility to use your own damage equation subroutine is provided. This option requires the availability of a FORTRAN or C/C++ compiler and linker.

You need to rebuild the `E2_UserSubroutine.dll` (Dynamic Linked Library) located in the `release_dlls` directory of the installation, as shown in the figure below.

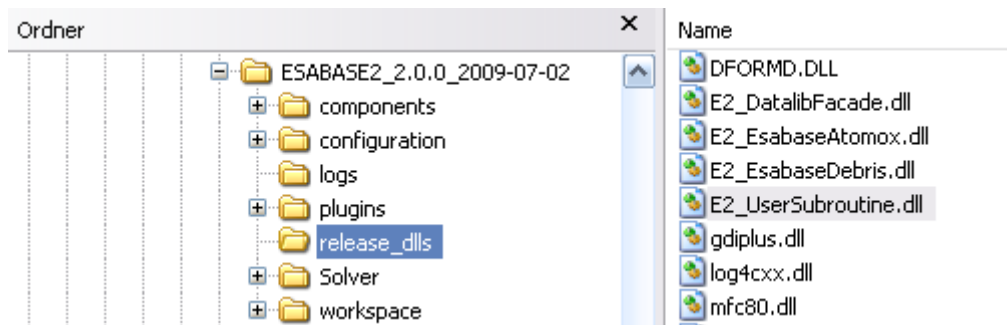


Figure 2.22: Finding the User subroutine DLL

ESABASE2 2.0.0 uses the Compaq Visual Fortran 6 compiler. It is highly advisable to use the same compiler. Also, while it is possible to use other languages (e.g. C++) to produce the subroutine DLL, compiler issues frequently occur.

Please note: When writing your own user subroutine in Fortran, take care to initialise all variables; otherwise, results can be erratic.

As a starting point, look at the Fortran file "user_dam.f" in the Debris plugin directory (`plugins\eu.esa.estec.esabase2005.debris_2.0.0\user_subroutine`). A simple example file is shown in Figure 2.23.

```

c---<kdb, etamax space>-----<2007-09-14>----
C
  SUBROUTINE USER_DAM (KFLAG, VPART, DPART, ALFA, RHOP, TB, RHOB,
&                      TS, RHOS, SPACE, X_OUT, D_USREQ)
C-----
C
C   Parameters of the calling sequence:
C
C   Input:
C     KFLAG: Calling Flag of the Routine :
C           1 - Routine called as ballistic limit equation
C           2 - Routine called as crater / hole size equation
C     VPART: Scalar impact velocity [km/s]
C     DPART: Particle diameter (for call as crater / hole size equation)
C     ALFA : Impact angle [deg]
C     RHOP : Particle density [g/cm3]
C     TB   : Back-up or Main Wall thickness [cm]
C     RHOB : Back-up or Main Wall density [g/cm3]
C     TS   : Shield cumulated thickness [cm]
C     RHOS : Shield average density [g/cm3]
C     SPACE: Cumulated spacing (shield to Back-up wall) [cm]
C
C   Output:
C     X_OUT: Routine output
C           Ballistic limit call: Particle limit diameter
C           Crater / hole size call: Crater / hole diameter
C
C   Input/Output:
C     D_USREQ: Parameters (up to 14) of the user failure and damage laws.
C-----
C   Purpose : Specification of any user defined failure (ballistic
C           limit) and damage (crater size) equation.
C           +++ This is a simple example routine. +++
C-----
C   CALLS : any user defined subroutine; to be included in
C           the UserSubroutine.dll
C-----
C   CALLED BY : DACRAHO, SH_BALDIV, RPRT_IMPA
C-----
C   COMMONS : none
C-----
C ... Input/Output Variable declaration
  INTEGER KFLAG
  REAL VPART, DPART, ALFA, RHOP, TB, RHOB, TS, RHOS, SPACE,
&      X_OUT
  REAL D_USREQ (14)
C ... USER_DAM internal variables
  REAL VNORM, ! Normal impact velocity
&      FMAX , ! Ballistic limit normalised to Dp**Lambda
&      XDUM  ! Dummy variable
C +++ Specification of d_usreq(1:14)
  D_USREQ(1) = 1.0d0
  D_USREQ(2) = 0.33d0
  D_USREQ(3) = 2.0d0
  D_USREQ(4) = 0.667d0
  D_USREQ(5) = 1.0d0
  D_USREQ(6) = 0.33d0
  D_USREQ(7) = 2.0d0
  D_USREQ(8) = 0.667d0
  D_USREQ(9) = 1.0d0
  D_USREQ(10) = 0.33d0
  D_USREQ(11) = 2.0d0
  D_USREQ(12) = 0.667d0
  D_USREQ(13) = 1.0d0
  D_USREQ(14) = 0.33d0

  VNORM = VPART * COS(ALFA)

  IF (KFLAG.EQ.2) THEN
C +++ Crater size equation
    XDUM = (DPART**1.056)*(RHOP**0.519)*(VNORM**0.66667)
    X_OUT = D_USREQ(11) * D_USREQ(12) * XDUM
  ELSE
C +++ Ballistic limit equation
    FMAX = D_USREQ(13)*D_USREQ(14) * (RHOP**0.519)*(VNORM**0.66667)
    X_OUT = (TB / FMAX)**(0.947)
  END IF

  RETURN
  END
C
C-----eof-----eof-----
C

```

Figure 2.23: Example user subroutine

After the variables specification, a block containing 14 parameters can be defined for the user damage equation.

The user sub-routine must be named `user_dam.f`.

The flag `KFLAG` indicates how the subroutine is called. When `user_dam` is called with `KFLAG = 1`, it is to be used as a ballistic limit equation, if `KFLAG = 2`, it is to be used as a crater / hole equation. Thus one subroutine can cover both types of equation.

It is possible to add further subroutines in the file `user_dam.f` which are called by the main user subroutine `USER_DAM`. Thus, the advanced user can build his own library of damage equations.

In case of persisting issues, please contact the ESABASE2 development team using the email address provided on the website /3/.

To apply the subroutine to the entire model, choose "User Subroutine" as damage equation in the Debris input editor, as described above (subsection 2.2.1.4). To apply it only to parts of the S/C geometry, go to the geometry editor, modify a shape via wizard, and on the Debris page, choose "User Subroutine" as shield type (see also section 2.1).

2.2.2 Debris Ground Test Tab

At the bottom of the Debris input editor, the second tab leads to the "Ground Test" page, which is depicted in the figure below.

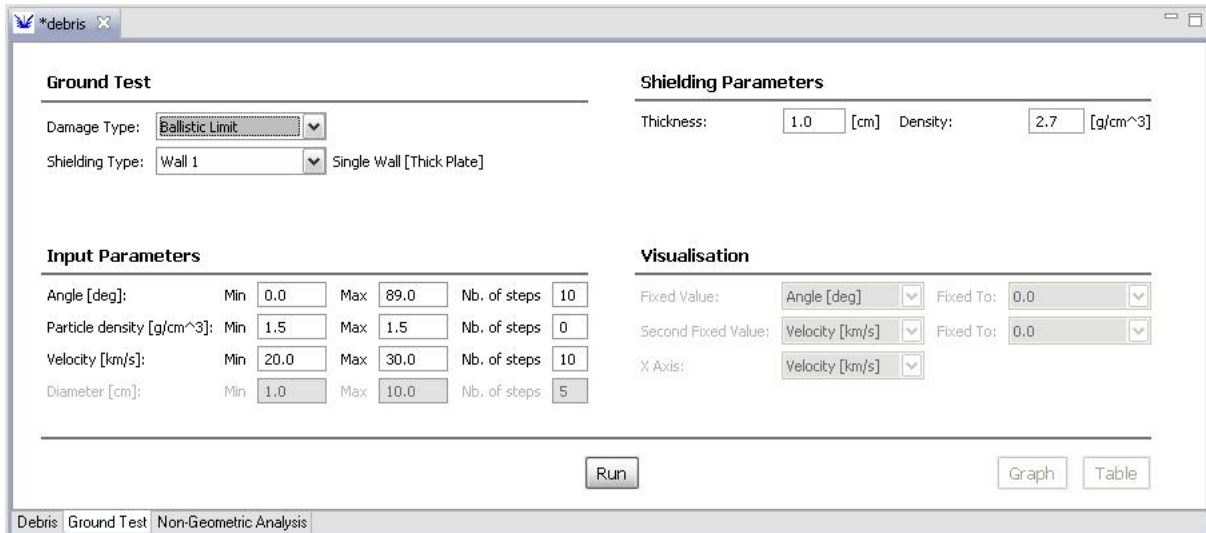


Figure 2.24: Debris input editor, Ground Test tab

The ground test option enables you to run the damage equations on their own, outside of the ESABASE2/Debris analysis; its purpose is to test and preview the results of the damage equations.

The following sections are available:

- *Ground Test:* Choose damage type (ballistic limit or crater size) and shielding type (Wall_1 to Wall_15). To define the wall configuration see section 2.2.1.4.
- *Shielding Parameters:* Choose shield thickness and density (and spacing in case of double walls).
- *Input Parameters:* Impact angle, density, velocity and diameter can be specified either as single values or as arrays of values. Each variable has three parameters: the minimum value, the maximum value and the number of steps. For single shots, only the minimum parameter is used. For tabled data, when the variable is used as x-axis or y-curve parameter, all three parameters are used.
- *Visualisation:* Choose the axis on the result graph.

Unlike the normal Debris analyses, the ground test can be performed directly within the Debris input editor; for this purpose, press the "Run" button. Afterwards, the "Graph" and "Table" buttons at the bottom right are enabled, leading to the popup windows illustrated in the screenshot below.

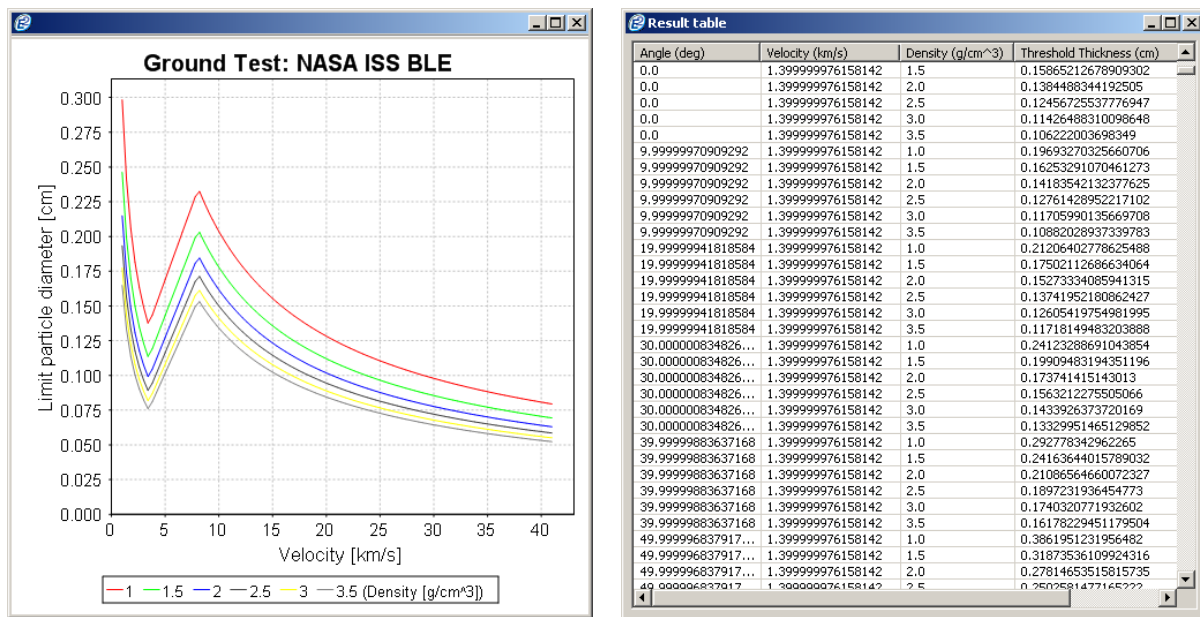


Figure 2.25: Debris input editor, Ground Test tab, Run

Graph and table show the same results.

2.2.3 Debris Non-Geometric Analysis Tab

The "non-geometric" analysis uses simplified spacecraft geometry: A pointed or randomly tumbling plate. This mode is used to check an analysis environment consisting of the orbit specification, the environment models and the damage assessment parameters.

Using this mode is recommended for large spacecraft models with many elements, in order to iron out input errors without running a time-consuming full analysis. The ray tracing scheme is the same for both geometric and non-geometric analyses.

The following figure shows the Non-Geometric Analysis tab in the Debris input editor.

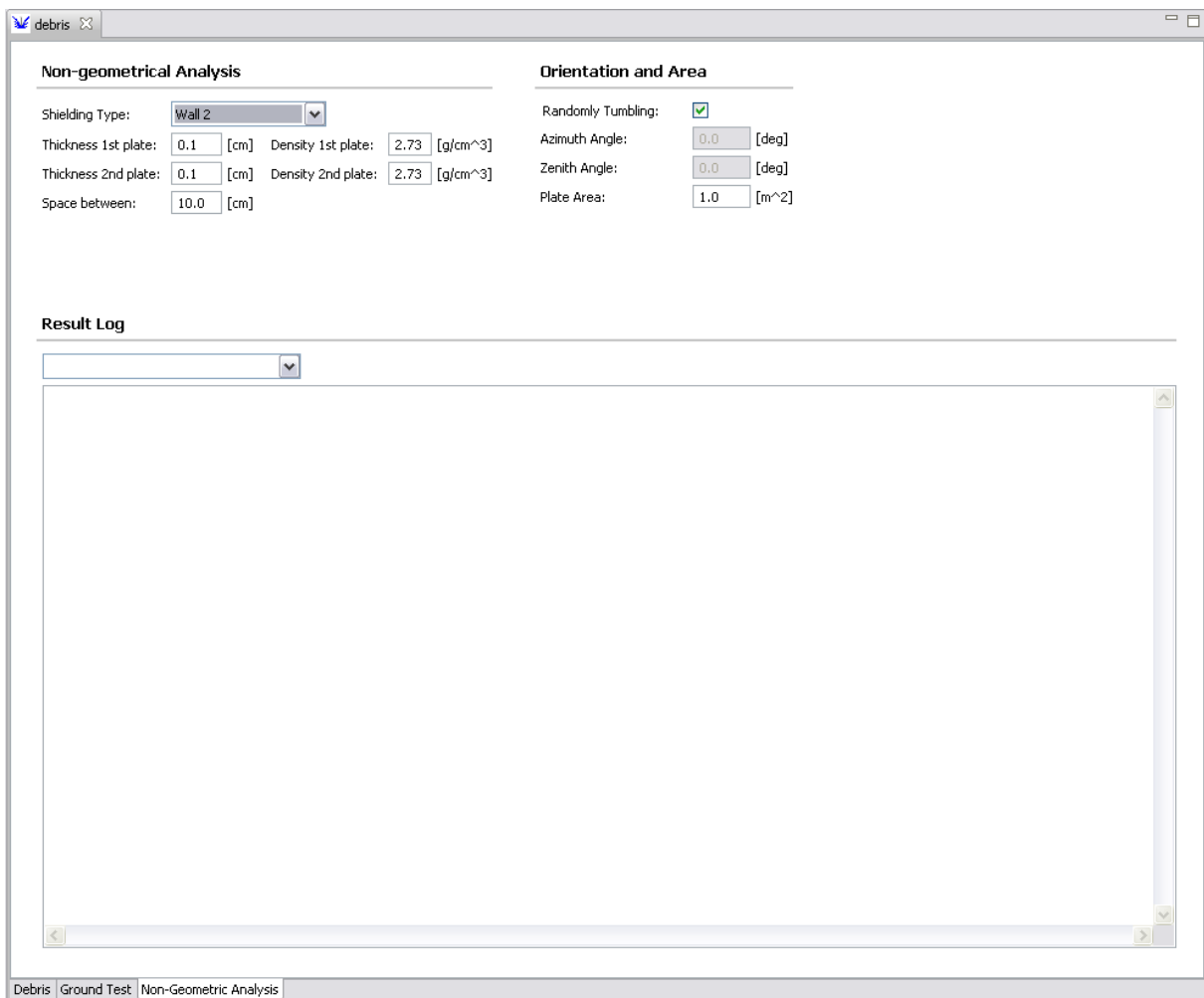


Figure 2.26: Debris input editor, Non-Geometric Analysis tab

In the tab, three sections are visible:

- *Non-geometrical Analysis:* Specifies shielding type (Wall_1 to Wall_15) as well as shield thickness and density (and spacing in case of double walls). To define the wall configuration see section 2.2.1.4.

- *Orientation and area*: Specifies the plate size and its orientation or randomly tumbling property.
- *Result Log*: Execution log of the analysis run (former listing file).

2.2.3.1 Non-Geometrical Analysis

The following screenshot shows the Non-Geometrical Analysis section for single (left) and multi wall (right) selection.

<p>Non-geometrical Analysis</p> <hr/> <p>Shielding Type: <input type="text" value="Wall 1"/> ▼</p> <p>Thickness: <input type="text" value="0.1"/> [cm] Density: <input type="text" value="2.7"/> [g/cm³]</p>	<p>Non-geometrical Analysis</p> <hr/> <p>Shielding Type: <input type="text" value="Wall 2"/> ▼</p> <p>Thickness 1st plate: <input type="text" value="0.1"/> [cm] Density 1st plate: <input type="text" value="2.7"/> [g/cm³]</p> <p>Thickness 2nd plate: <input type="text" value="0.1"/> [cm] Density 2nd plate: <input type="text" value="2.7"/> [g/cm³]</p> <p>Space between: <input type="text" value="10.0"/> [cm]</p>
---	--

Figure 2.27: Debris input editor, Non-Geometric Analysis tab, Single and Multi wall

The input values correspond to the ones used in the geometry editor, in the Debris page of the shape wizards (see 2.1).

2.2.3.2 Plate Orientation and Area

The figure below shows the Plate Orientation and Area section within the Non-Geometric Analysis tab.

<p>Orientation and Area</p> <hr/> <p>:Randomly Tumbling <input type="checkbox"/></p> <p>Azimuth Angle: <input type="text" value="0.0"/> [deg]</p> <p>Zenit Angle: <input type="text" value="0.0"/> [deg]</p> <p>Plate Area: <input type="text" value="1.0"/> [m²]</p>	<p>Orientation and Area</p> <hr/> <p>:Randomly Tumbling <input checked="" type="checkbox"/></p> <p>Azimuth Angle: <input type="text" value="0.0"/> [deg]</p> <p>Zenit Angle: <input type="text" value="0.0"/> [deg]</p> <p>Plate Area: <input type="text" value="1.0"/> [m²]</p>
---	--

Figure 2.28: Debris input editor, Non-Geometric Analysis tab, Orientation and Area

The plate orientation can be one of the following.

- *Randomly tumbling*: This is the random tumbling plate (RTP) mode. It often corresponds to the environment models themselves, and is an effective way to get a quick first-order assessment of the micro-particle environment risk for a mission.
- *Oriented*: For this mode, azimuth and zenith angles of the plate normal can be defined.
 - The azimuth angle is the angle with respect to the velocity direction (0 deg corresponds to the ram direction, positive towards right).
 - The zenith angle is the angle to the space (zenith) direction (0 deg corresponds to the space direction).

In both cases, the area of the plate can be defined. Normally the default (1 m²) is used, but a specific area can be specified instead, e.g. in case of a detector or other special surface.

2.2.3.3 Result Log

The results of a non-geometrical analysis are displayed in the bottom part of the Debris input editor. You can select between four different sets of results:

- hits vs. crater size
- debris and meteoroid flux and damage results
- failures vs. ballistic limit
- orbit propagation results

All results will be presented in textual form (ASCII files).

The results listings are also available in the `.\ListingFiles` folder of the corresponding project directory. The names of the listing files are composed from the string 'NonGeom', the analysis date and time and the type of listing (result set).

2.3 Debris Analysis

With both debris information within the spacecraft geometry and global debris input parameters specified, the time has come to perform a debris analysis. There are two types of analysis runs:

- *Geometric Analysis*: This is what you would normally expect: A mission and a spacecraft geometry are sent to orbit propagation and debris analysis (see section 2.3.1).

- *Non-geometric Analysis*: Subtracts the S/C geometry from the analysis, instead taking a simple plate (see section 2.3.2).

2.3.1 Geometric Debris Analysis

In the toolbar at the top of the application window, locate the "Run" button depicted in the following figure. Press the small down-arrow at the right of the button and a context menu will appear.

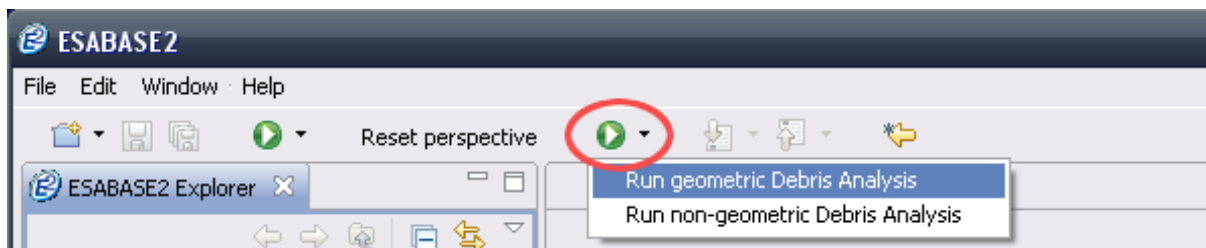


Figure 2.29: Geometric Debris analysis, Run button

Whenever you click directly on the Run button instead of the down-arrow to the right, the last selected run type will be repeated.

Choose the first entry, "Run geometric Debris Analysis". A wizard as illustrated by the following screenshot will appear.

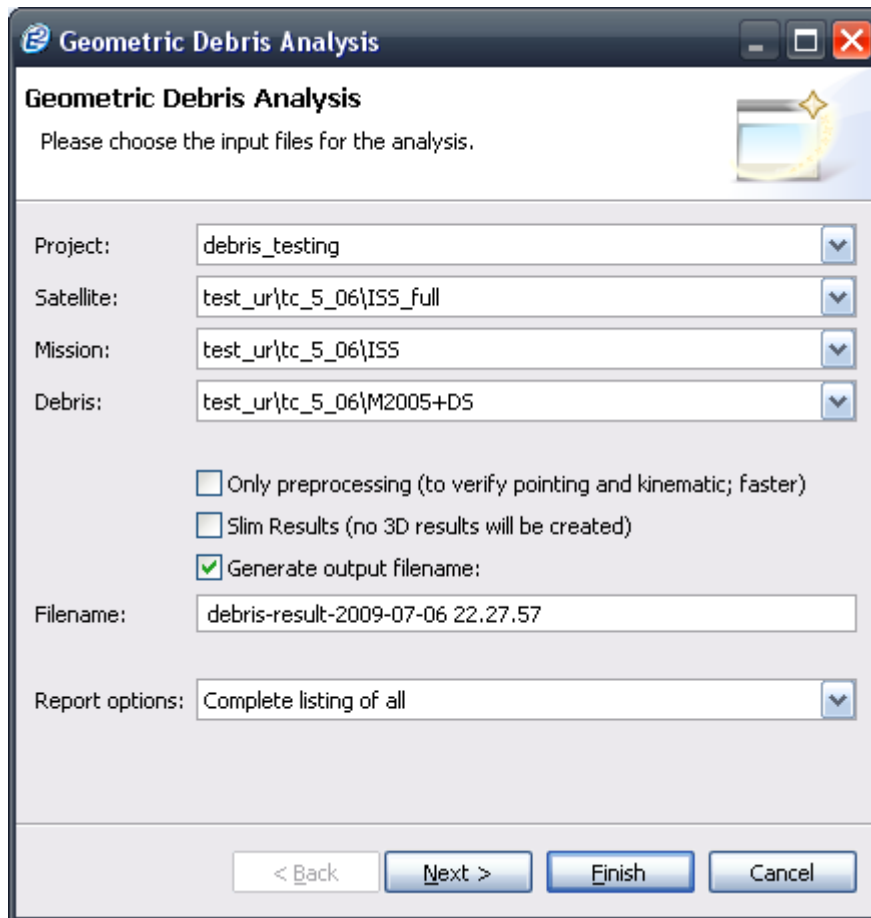


Figure 2.30: Geometric Debris analysis, Run wizard

In the first line, you specify a project from your workspace; then, the next three combo-boxes will only show files from this project. A spacecraft geometry, a mission file and a debris input file must be chosen.

Activating the "Only Preprocessing" checkbox will stop the analysis after pointing and kinematics have been calculated. It is useful to check the correctness of the pointing and kinematics settings without running a complete and possibly time-consuming debris or meteoroid analysis.

The "Slim Results" checkbox is a performance switch. With it, LIS* files (as in the original ESABASE) and 2D graphs will be created as usual and with full information content (as specified in the "Report Options" box), but the only 3D geometry result will be for the entire mission; no orbital points or arcs will be available. For large models or many orbital points, this may be the only option to run ESABASE2/Debris within the available RAM of the executing computer.

Generating an output filename is highly recommended. The default filename contains the solver type (here: "debris") and the current date and time.

At the bottom, the "Report options" combo-box allows you to specify the content of the "Deb Met" listing (formerly *.LISD, *.LISM, *.LISDM output files) within the Debris result file. You can choose between the following options:

- *Output of a summary of objects:* results on orbital arc and mission level, and on object and spacecraft level (no object/element summary, no orbital point related results and no element related results).
- *Output of header and summary all:* the same, but including object/element summary and element related results.
- *Complete listing of objects:* orbital point, orbital arc and mission related results on object and spacecraft level (no object/element summary, and no element related results).
- *Complete listing of all:* the same, but including object/element summary and element related results.

Optionally, you can press the "Next" button to go to the second page of the Debris analysis wizard, which is depicted in the figure below.

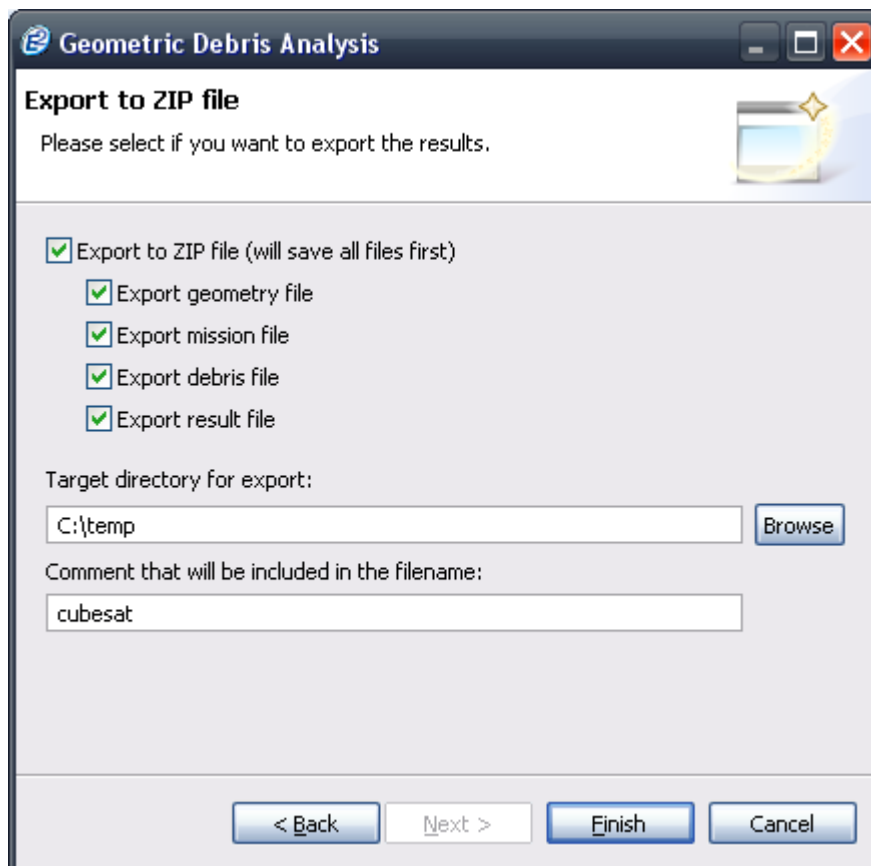


Figure 2.31: Geometric Debris analysis, Run dialog, export page

The debris result file will be saved into your workspace. This page allows you to additionally export the input and result files to a ZIP file suitable for transmission via email or other means.

Press the "Finish" button to launch the Debris analysis.

A progress bar will appear and keep you updated about the current state of the application. Depending on the number of elements in the S/C geometry and depending on the number of orbital points, an analysis might take a long time (i.e. several hours).

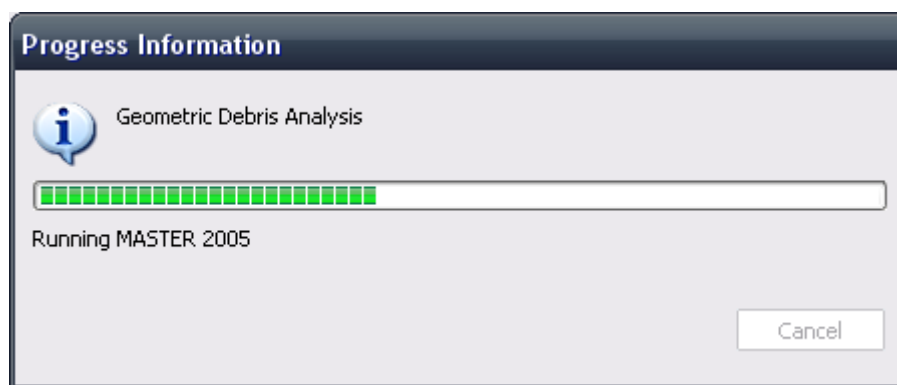


Figure 2.32: Geometric Debris analysis, Progress bar

After the run, a debris result file will be created in your workspace, and it will be automatically opened. If you are interested in its contents, skip to section 2.4; otherwise, the following subsection will explain the non-geometric debris analysis.

2.3.2 Non-geometric Debris Analysis

The non-geometric analysis uses a simple plate instead of a full spacecraft geometry; it is therefore considerably faster. As in the geometric debris analysis, locate the Run button and, this time, choose "Run non-geometric Debris Analysis".

The following figure shows the Run wizard for the non-geometric debris analysis.

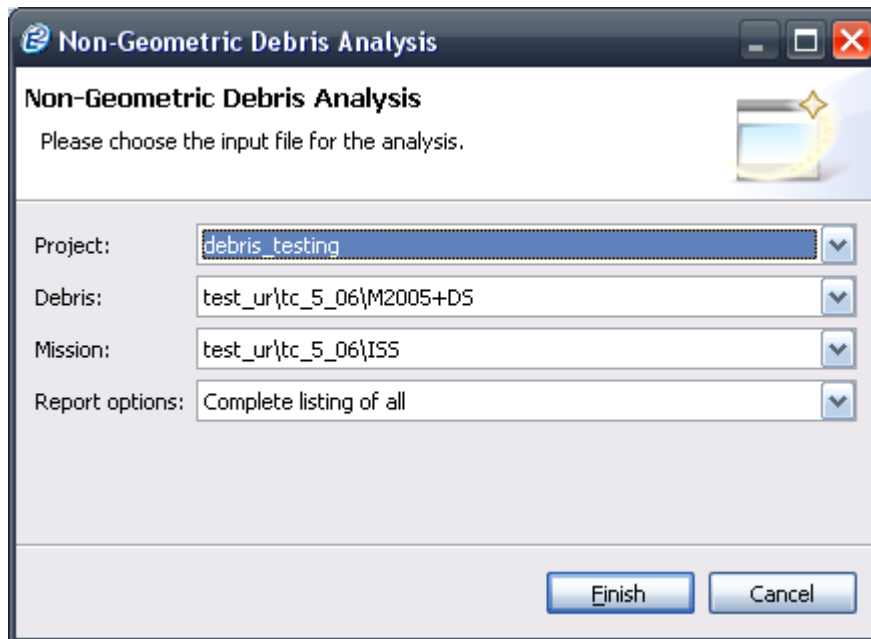


Figure 2.33: Non-geometric Debris analysis, Run wizard

Compared to the geometric Debris analysis, this wizard is considerably simpler. The spacecraft geometry input file is omitted and all options have been removed; only the report options remain.

No output file needs to be specified since the result listings are displayed in the "Result Log" section of the Debris input editor's "Non Geometric Analysis" tab. The content of the listings is stored within the debris input file.

The screenshot below shows the debris input editor, supplied with results from a non-geometrical debris analysis.

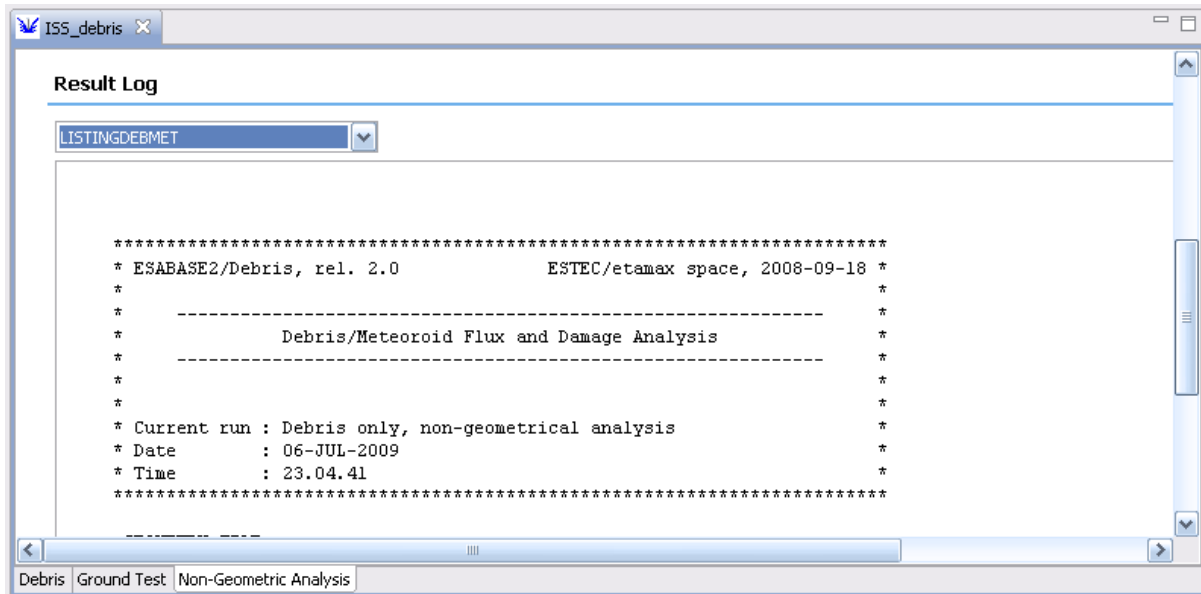


Figure 2.34: Non-geometric Debris analysis, results

In the "Result Log" section, the combo-box toggles between the output files, while the text area below shows the content of the selected file. Here, the Debris/Meteoroid listing ("LISTINGDEBMET") is shown.

2.4 Debris Results

A geometric Debris analysis produces a debris result file, which is interpreted by the Debris result editor. The editor contains the following functionalities:

- *3D Results*: Shows the spacecraft geometry overlaid with the impact flux or other output parameters (see section 2.4.1).
- *2D Results*: Graphs with different data about flux distributions at the orbital points as well as average flux distributions (see section 2.4.2).
- *Listings*: In tradition of the original ESABASE, this tab shows the LIS* files produced by the Debris solver (see section 2.4.3).
- *Notes*: A blank text area for your own notes (see section 2.4.4).

2.4.1 3D Results

The following figure shows the "3D Results" tab in a debris result editor.

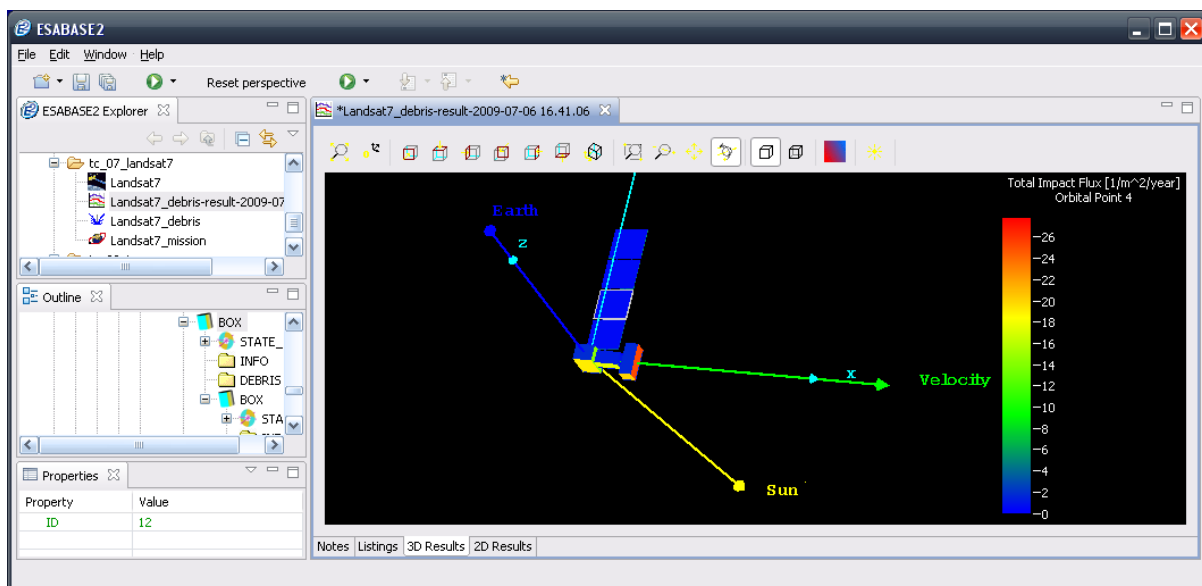


Figure 2.35: Debris result editor, 3D Results

You can see a spacecraft geometry (Landsat7) with total impact flux, visualised by colour codes on the elements and decoded by the colour scale on the right. At the top of the editor, the same toolbar as in the geometry editor appears, allowing you to zoom, rotate and scroll around the spacecraft.

Additionally, you can rightclick into the 3D area to invoke the context menu, which contains the following options:

- *Colour*: Defines the result value to be laid over the S/C geometry; default is total impact flux.
- *Orbital Point*: Whether to show results for the entire mission, single orbital points or an orbital arc.
- *Coordinate Systems*: Whether to display coordinate systems alongside the S/C geometry.
- *Element Chart*: Shows the results for one single geometrical element (select one with Shift+Leftclick to enable).

2.4.1.1 Colour

Debris results come in multiple variables, e.g. impact flux or failure flux (and either per year or over the entire mission time). Only one of these variables at a time can be laid over the geometry model.

To choose a result set to be displayed, rightclick the 3D view to open the context menu, then choose "Colour → <Variable>", as shown in the following figure.

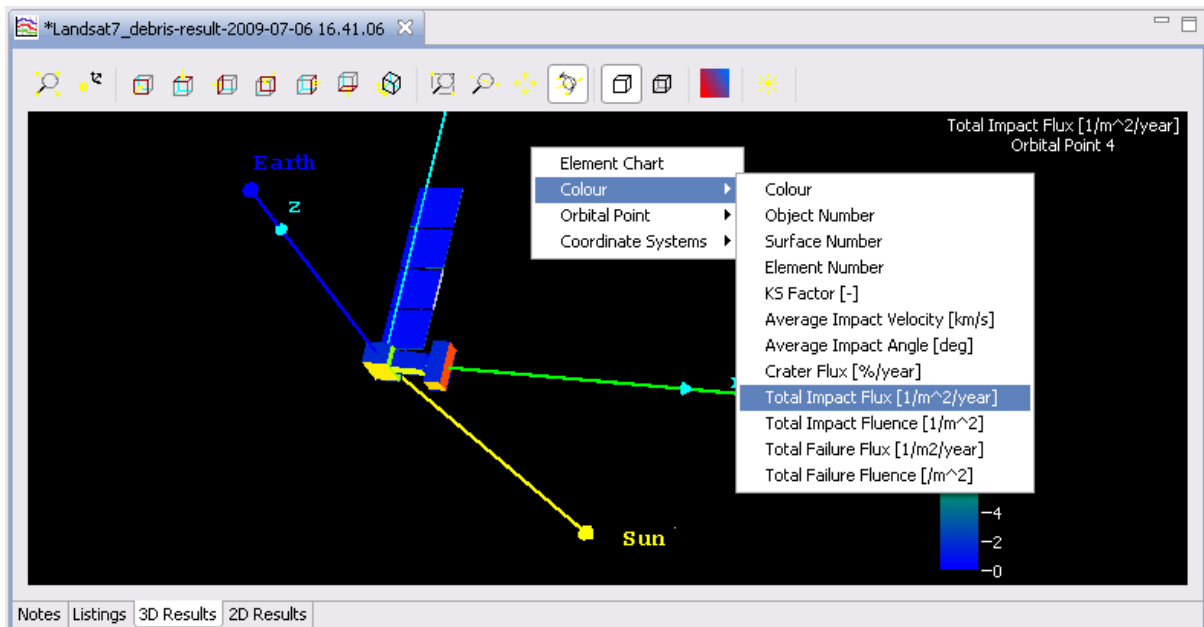


Figure 2.36: Debris result editor, 3D Results, colour context menu

The first 4 items (listed below) are not part of the Debris results; instead they are used to gain an overview of the spacecraft geometry, and particular its objects, surfaces and elements.

- *Colour*: Displays exactly the same object colour which was used in the geometry editor.
- *Object Number*: Each object is identified by an object number. In this overlay, this object number is mapped to colours.
 - Leftclick on an object in the 3D view to select it, and then look at the colour scale to the right. The appropriate colour will be marked, and the object number will be shown.
- *Surface Number*: Breaking down the objects, each surface is shown in a different colour.
 - Ctrl+leftclick on a surface in the 3D view to select it. The respective surface number is marked in the colour scale to the right.
- *Element Number*: Further breaking down the surfaces, each element is shown in a different colour.
 - Shift+leftclick on an element in the 3D view. The selected element number and color will be marked in the color scale.

The following 8 items show the respective quantity (listed below) as colours on the elements of the geometry model. To see the exact value on an element, Shift+leftclick the element; the colour scale on the right will then show the quantity value.

- KS Factor
- Average Impact Velocity [km/s]
- Average Impact Angle [deg]
- Crater Flux [%/year]
- Total Impact Flux [$1/m^2/year$]
- Total Impact Fluence [$1/m^2$]
- Total Failure Flux [$1/m^2/year$]
- Total Failure Fluence [$1/m^2$]

If the secondary ejecta option was activated in the Debris input editor, it is now additionally possible to display the direct impact/failure flux/fluence and the ejecta impact/failure flux/fluence.

Please note that results always sum up both sides of all elements. A distinction between positive and negative side of an element is not possible.

2.4.1.2 Orbital Point

When the debris result editor opens, it shows the analysis results for the entire mission. You can also view the results for each orbital point or for the orbital arc. Open the context menu and navigate to "Orbital Point", as shown in the following figure.

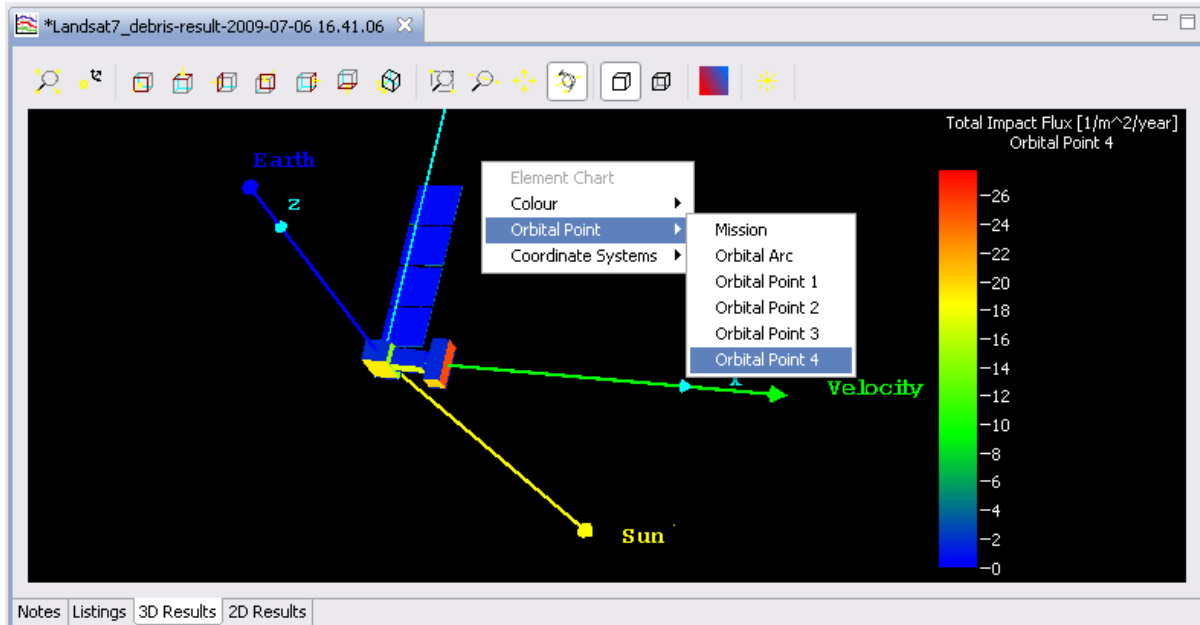


Figure 2.37: Debris result editor, 3D Results, Orbital Point context menu

Choose one of the following:

- *Mission*: The entire mission duration.
- *Orbital Arc*: One orbital arc only.
- *One of the orbital points*: Results at a dedicated orbital point.

When you select an orbital point, then Earth, Sun and velocity direction are shown as depicted above. Note that the correct pointing (e.g. solar panel aligned towards the Sun) can only be shown at the orbital points, not for the whole mission or orbital arc.

2.4.1.3 Coordinate Systems

To visualise the coordinate system you used in the geometry editor, change the coordinate system by opening the context menu and navigating to "Coordinate Systems", as shown in the following figure.

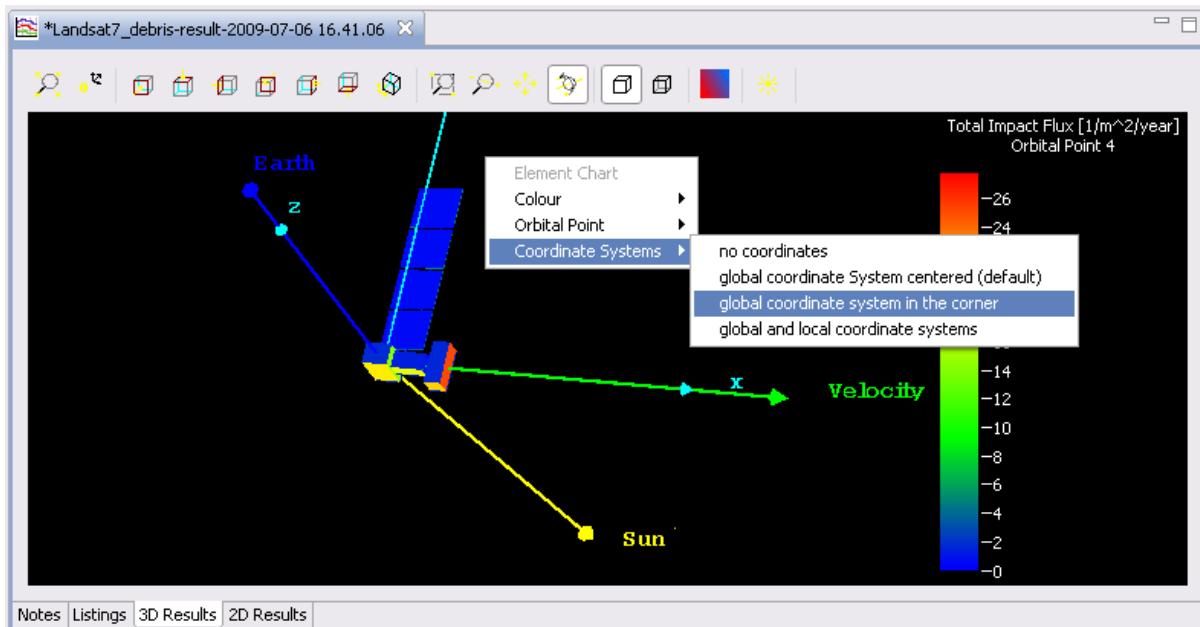


Figure 2.38: Debris result editor, 3D Results, Coordinate Systems context menu

You can choose among the following coordinate system settings:

- *No coordinates:* Note that pointing vectors will also be deactivated.
- *Global coordinate system (centred):* Shows the xyz-axis for the entire system.
- *Global coordinate system in the corner:* Same as before, but in the corner of the system, not centred.
- *Global and local coordinate systems:* Shows xyz-axis for each object. Detailed but probably slightly irritating for complex S/C geometries.

2.4.1.4 Element Chart

Different 2D graphs of the debris results can be shown for the entire spacecraft geometry using the "2D Charts" tab, as described in the next section (2.4.2). However it is also possible to show graphs of only one element.

Shift+Leftclick an element of the S/C geometry, then open the context menu and select "Element Chart". The following figure shows the resulting small popup window.

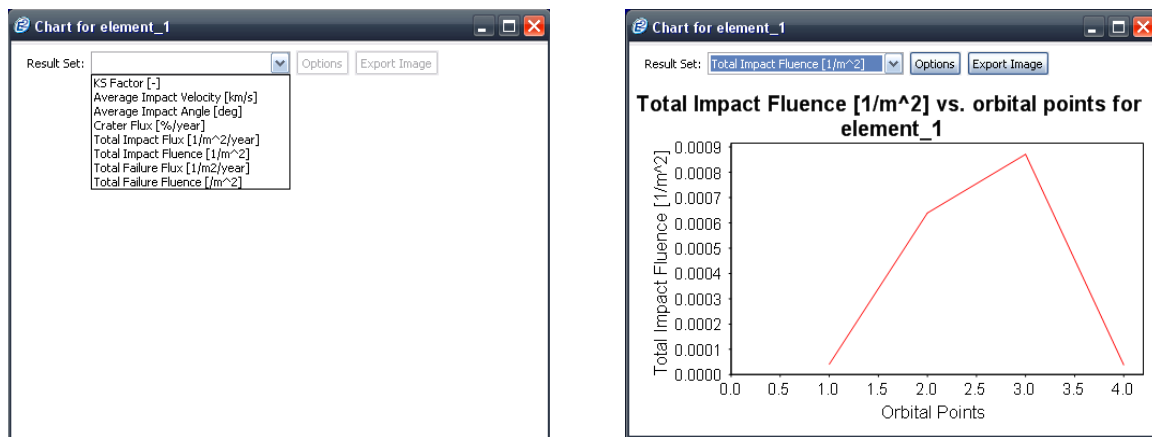


Figure 2.39: Debris result editor, 3D Results, Element Chart

From the "Result Set" combo-box, select a variable; the appropriate graph for the given element will be displayed, as shown on the right. These are the same values that are colour coded in the element in the 3D view.

You also have the possibility to adjust the appearance of the graph via the "Options" button or a context menu opening on rightclicking, and to export the image as PNG or JPG via the "Export Image" button (s. sections 2.4.2.1 and 2.4.2.2 for details).

2.4.2 2D Results

Complementing the "3D Results" tab described above, charts can show different flux distributions in the "2D Results" tab. An example is shown in the following figure:

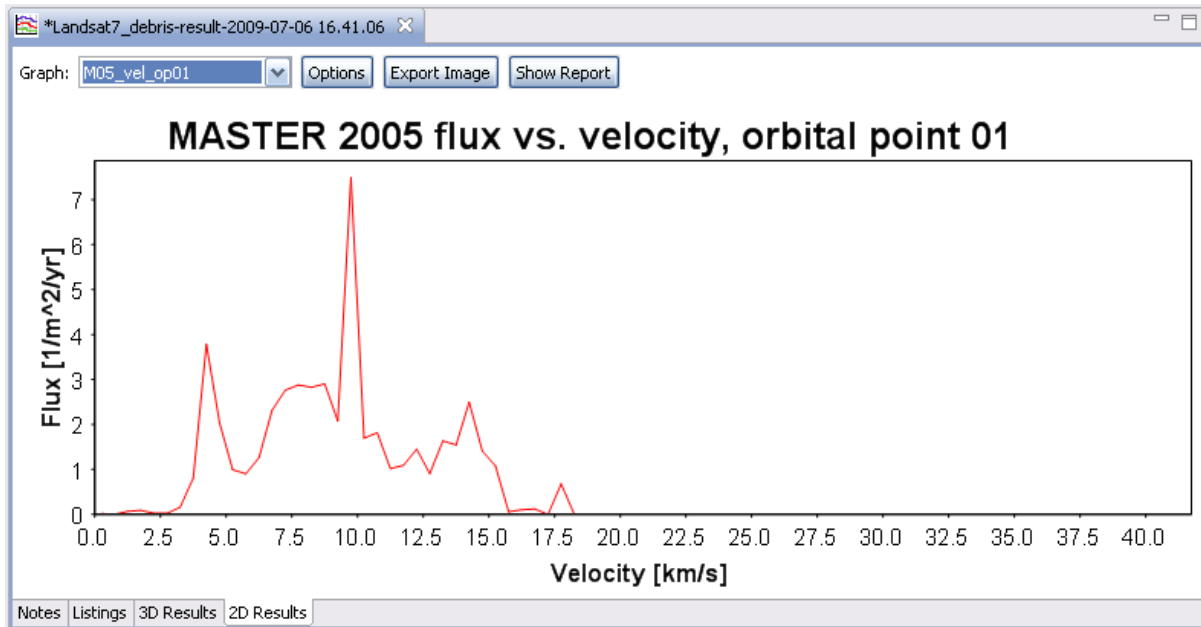


Figure 2.40: Debris result editor, 2D Results

In the combo-box on the top left, you can select the chart you want to see. Which charts are available depends on the debris/meteoroid model used for the analysis, as shown in the following table.

Debris / meteoroid model	Flux distributions
NASA90	$F = f(d)$ (valid for all orbital points)
MASTER 2001	$F = f(d)$, $F = f(\text{azimuth})$, $F = f(\text{elevation})$, $F = f(\text{velocity})$ for each orbital point
MASTER 2005	$F = f(d)$, $F = f(\text{azimuth})$, $F = f(\text{elevation})$, $F = f(\text{velocity})$ for each orbital point
MASTER 2009	$F = f(d)$, $F = f(\text{azimuth})$, $F = f(\text{elevation})$, $F = f(\text{velocity})$ for each orbital point
ORDEM2000	$F = f(d)$, $F = f(\text{azimuth})$ for each orbital point

Debris / meteoroid model	Flux distributions
Grün	$F = f(m)$ (valid for all orbital points)
Divine-Staubach	$F = f(d)$, $F = f(\text{azimuth})$, $F = f(\text{elevation})$ for each orbital point
MEM	$F = f(m)$, $F = f(\text{azimuth})$, $F = f(\text{elevation})$, $F = f(\text{velocity})$ for each orbital point
LunarMEM	$F = f(m)$, $F = f(\text{azimuth})$, $F = f(\text{elevation})$, $F = f(\text{velocity})$ for each orbital point

Table 2.1: Available 2D charts in dependency of the Debris/Meteoroid model

The "2D Result" tab offers additional functionality to work with the charts, represented by buttons to the right of the chart combo-box:

- *Options*: Allows you to customise the chart appearance.
- *Image Export*: Save the current chart to a PNG or JPG file.
- *Show Report*: Opens a print-ready report for a given chart.

2.4.2.1 2D Charts: Options

The "Options" button opens a context menu for a given chart, as shown in the following figure. It contains axis configurations as well as settings for the data lines.

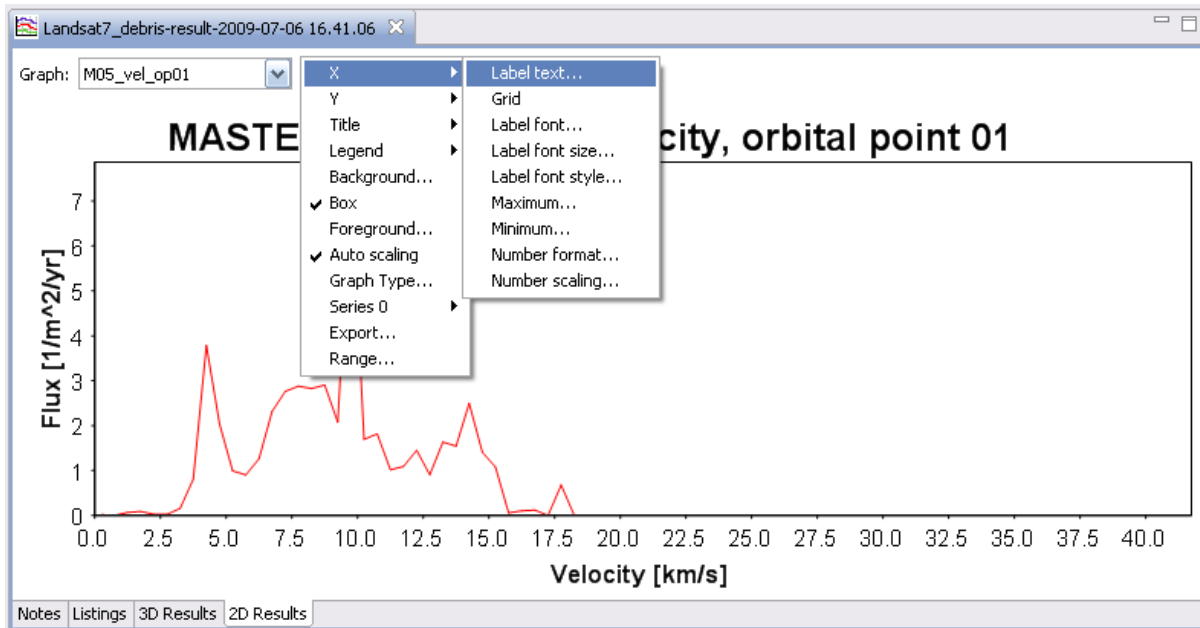


Figure 2.41: Debris result editor, 2D Results, options

You can customize the following chart properties:

- x, y axis
 - *Label Text:* Changes the text of the axis label.
 - *Grid:* Displays horizontal/vertical dashed lines.
 - *Label font, Label font size, Label font style:* Changes the appearance of the axis label.
 - *Maximum, Minimum:* The value range that this chart shows
 - *Number format:* The format of the number labels, e.g. the number of decimal places, can be specified.
 - *Number scaling:* Choose between linear or logarithmic scale.
- Title
 - *Graph title:* The title text above the chart.
 - *Title visible:* Show or hide the title text.
 - *Title Font, Title Font Size, Title Font Style:* The appearance of the title text.
- Legend
 - *Legend:* Show or hide the legend.
 - *Legend Font, Legend Font Size, Legend Font Style:* The appearance of the legend.
- *Background:* The background color of the chart.
- *Box:* Shows or hides the box around the graph.

- *Axis and label colour*: The colour of the axis lines and the labels.
- *Major tick labels*: Shows or hides number labels at the major ticks.
- *Minor tick marks*: Display small ticks between the major ticks.
- *Ratio*: Set a fixed aspect ratio.
- *Auto scaling*: Re-scales the chart.
- *Graph Type*: Choose between line-, bar- and scatter chart.
- Series <num>:
 - *Linecolour, Linewidth*: Color and width of the line of this data series.
 - *Stroke*: Type of the line of this data series.
- *Range*: Set the X and Y range that the chart covers.

2.4.2.2 2D Charts: Image Export

Pressing the "Export Image" button opens a file dialog asking for a location to save to. PNG and JPG format are possible. The size of the image is the same as it is currently shown; re-size the application window to get other image sizes.

2.4.2.3 2D Charts: Show Report

The chart can be embedded into a print-ready report; press the "Show Report" button to invoke it. After a dialog asking for a comment line has been confirmed, a report as shown in the following figure will appear in a popup window.

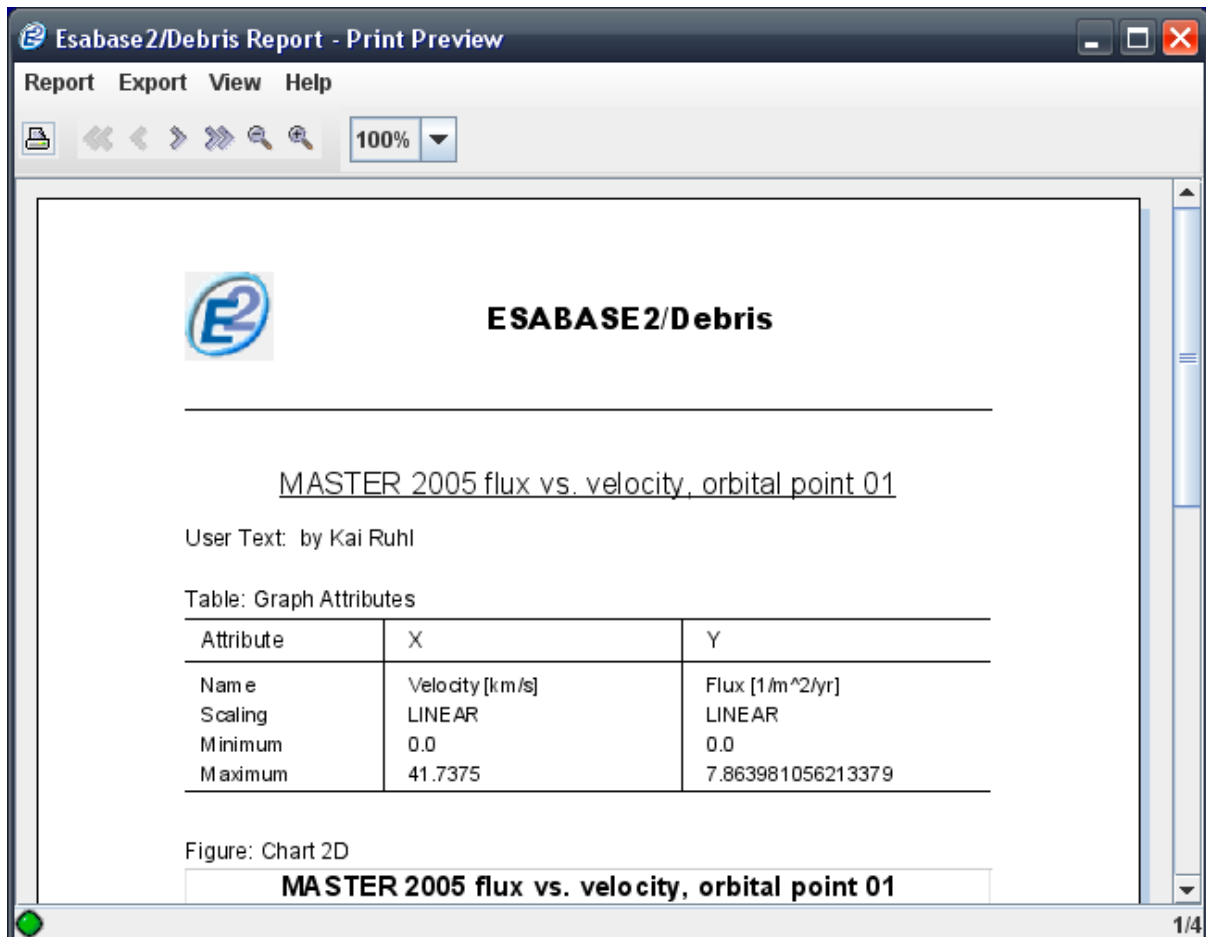


Figure 2.42: Debris result editor, 2D Results, reporting

This report details the chart properties and below the chart itself, followed by the data values represented in the chart (on the second and following pages). It is thus a complete data description of a value pair's relation, suitable for distribution to your colleagues.

You may either print the report (press the print button) or choose "Export → Save as PDF" for electronic distribution.

2.4.3 Listings

The original ESABASE produced LIS* files (or listing files), containing all debris results in one text file. This ability is also available in ESABASE2 in order to stay compatible to many post-processing tools.

The listings are shown on the "Listings" tab of the debris result editor, as shown in the figure below.

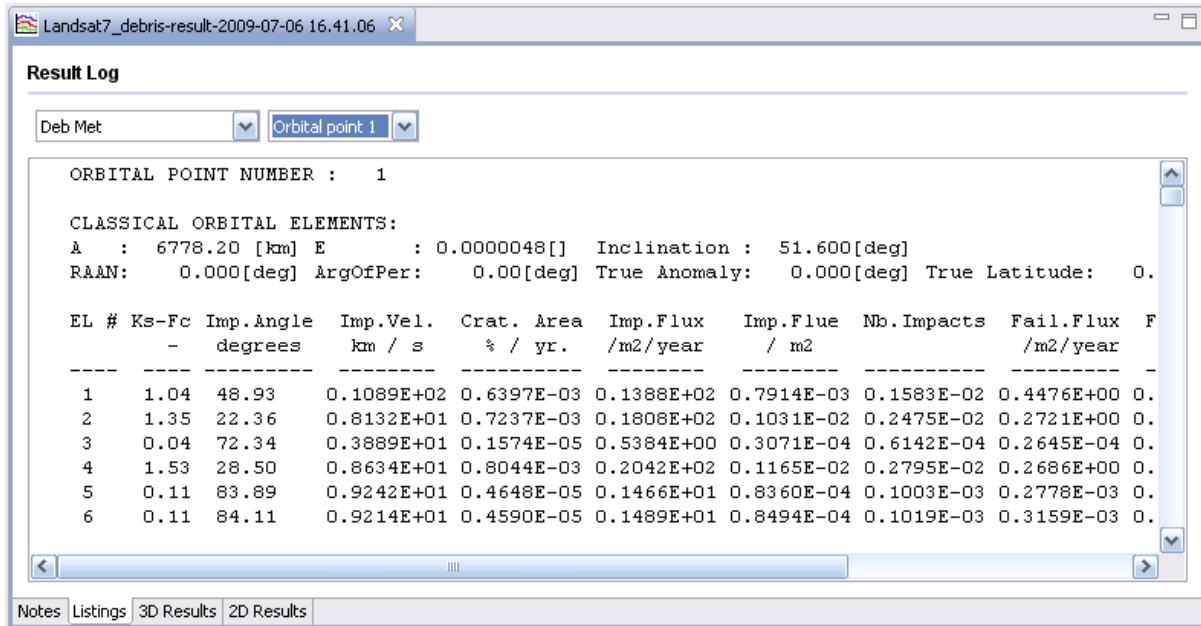


Figure 2.43: Debris result editor, 2D results, listings

In the result log, you can see the different LIS files; just select the appropriate LIS entry in the combo-box. For Debris/Meteoroid (Deb Met), you can jump to orbital point bookmarks using the second combo-box. The following LIS files are available:

- Crater vs. Crater size
- Deb Met (Debris/Meteoroid)
- Failures vs. Ballistic Limit
- Kinematic
- Orbit

Detailed information about the contents of the listing files can be found in the original ESABASE user manual /4/.

Please note: The listing files are foremost saved as data nodes in the debris result file. Additionally, they are written in ASCII format to the "Listings" subfolder of the current project folder, in order to be more easily accessible to post-processing tools.

Due to some confusion in the past the interpretation of the Crater vs. Crater size LIS file is described in the following section.

2.4.3.1 Crater vs. Crater size

The Figure 2.44 shows an example of a Crater vs. Crater size LIS file. The name indicates that it tabulates the number of craters again a crater size. The results are cumulative.

Result Log

Crater vs. Cratersize

HITS vs CRATER SIZE File K:\ESABASE2_temp\ESABASE2_6.0.0\workspace\Example\ListingFiles\Plate-Grun_CRT_D.CRT_D

IELMT	Elm Area	.1000E-02	.1000E-01	.1000E+00	.1000E+01	.1000E+02	.1000E+03	.1000E+04
1	0.723	.17946E+03	.15043E+03	.32011E+01	.25539E-02	.53327E-06	.00000E+00	.00000E+00
2	0.723	.17573E+03	.14689E+03	.31543E+01	.25253E-02	.52793E-06	.00000E+00	.00000E+00
3	0.015	.17687E+01	.12261E+01	.19729E-01	.14002E-04	.28367E-08	.00000E+00	.00000E+00
4	0.015	.17803E+01	.12288E+01	.19631E-01	.13774E-04	.27809E-08	.00000E+00	.00000E+00
5	0.017	.27159E+00	.65109E-01	.29004E-03	.10363E-06	.76045E-11	.00000E+00	.00000E+00
6	0.017	.26408E+00	.64745E-01	.31146E-03	.11697E-06	.11308E-10	.00000E+00	.00000E+00
7	0.015	.17665E+01	.12030E+01	.18494E-01	.12609E-04	.25269E-08	.00000E+00	.00000E+00
8	0.015	.17978E+01	.12311E+01	.19380E-01	.13432E-04	.27029E-08	.00000E+00	.00000E+00
9	0.017	.29001E+01	.19323E+01	.29708E-01	.20618E-04	.41490E-08	.00000E+00	.00000E+00
10	0.017	.29610E+01	.20053E+01	.31162E-01	.21612E-04	.43512E-08	.00000E+00	.00000E+00
11	0.723	.26833E+02	.13917E+02	.17356E+00	.11468E-03	.22471E-07	.00000E+00	.00000E+00
12	0.723	.27332E+02	.14160E+02	.18190E+00	.12225E-03	.24117E-07	.00000E+00	.00000E+00

Figure 2.44: Example of Crater vs. Crater size LIS file.

The first column "IELMT" lists the elements of the model, thus the results of the Crater vs. Crater size LIS file are on the element level. The corresponding results on the object level are presented on the bottom of the Deb Met LIS file. Both LIS files provide the mission level of hits.

The second column lists the areas of the corresponding elements in [m²].

The columns 3 to 9 provide results (number of hits) for seven different crater size thresholds. This means the listed hit causes a crater that is equal or larger than the threshold crater size. The first (minimum) threshold is the *Min. crater diameter* defined by the user, see 2.2.1.2. Based on this value the other six thresholds are defined by raising the crater size threshold by one order of magnitude. This can be seen very well in the first line of the table in Figure 2.44, which lists the values of the thresholds as head of the columns.

In the same way the LIS file Failures vs. Ballistic Limit can be interpreted. In this case the number of failures (instead of number of craters) is tabulated again the ballistic limit (instead of crater size) of the back wall. It uses the same principle and definition of the thresholds.

2.4.4 Notes

The figure below shows the "Notes" tab within the Debris result editor.

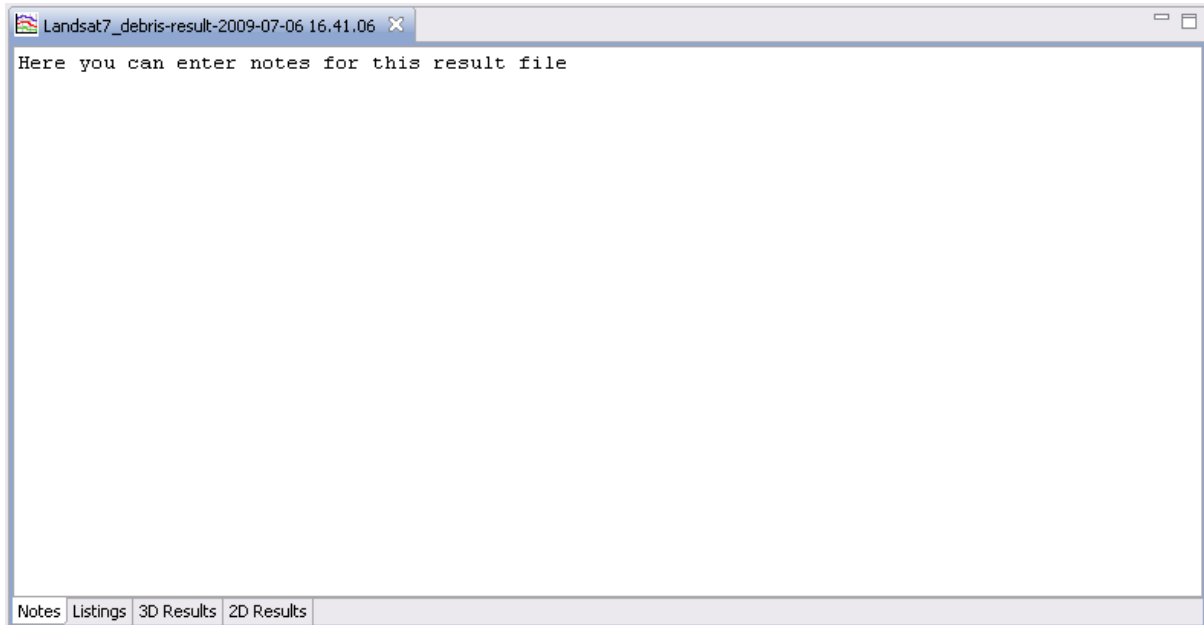


Figure 2.45: Debris result editor, 2D Results, notes

It consists of a simple text area, where you can write your notes concerning the debris results. Your text will be saved, and is available to whomever you may send the file.

3 Lunar Mission Debris Analysis

The intention of this chapter is to provide a small overview how a debris analysis for lunar mission can be performed.

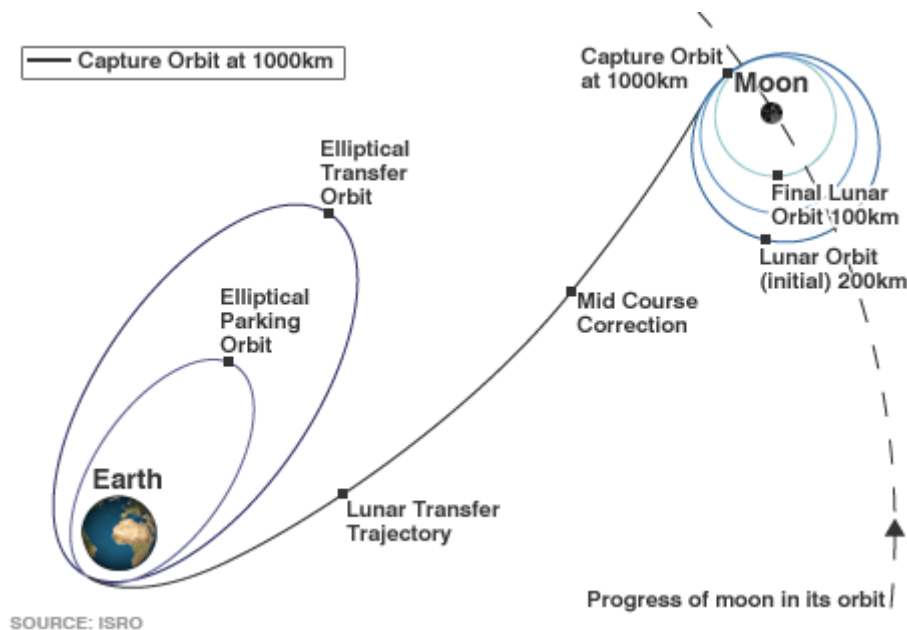


Figure 3.1: Scheme of the lunar mission of Chandrayaan 1 (source: North East India News)

The Figure 3.1 displays an example of a trajectory of a lunar mission, as was used for Chandrayaan 1.

3.1 Lunar Mission

Two different approaches can be used for an analysis of a lunar mission:

- Abstraction of the mission by splitting it into multiple orbits;
- Describing the mission with state vectors provided via a trajectory file.

In the first case the results of the individual orbits have to be post-processed manually to obtain the results for the whole mission.

The second case provides complete mission results. In this case, a trajectory file is used, describing the whole mission with up to 100 orbital points.

Both approaches for the mission definition are described in more detail in the following sections.

3.1.1 Abstraction of the Lunar Mission

A lunar mission can be abstracted by the definition of multiple orbits. The simplest case includes three orbits (only the journey from Earth to Moon):

- A low Earth orbit (LEO), which is often called the parking orbit;
- A highly elliptical orbit (HEO), the transfer orbit, with a perigee at the parking orbit and the apogee at the orbit of the Moon;
- A low lunar orbit (LLO), which is the target orbit of the mission.

Each of the orbits represents one of the phases of a lunar mission. The three phases describe roughly the lunar mission; the first phase includes the mission time in the vicinity of the Earth, here it is represented by a LEO. The second phase is the transfer from Earth to Moon, represented by a HEO. The third phase of the mission is the motion of the S/C in the vicinity of the Moon, in this case a LLO.

The individual orbits generated in the following give only an outlook of possible tracks and are not combined to reflect a real mission. To abstract a real mission the used mission epochs and durations have to fit a meaningful timeline, i.e. the mission epochs have to form a continuous time interval (second track starts at the end of the first track etc.). Furthermore the start epoch and orbit parameters of the transfer orbit have to be defined in a way that the Moon interferes with the transfer orbit and the S/C is at the according sector of the orbit at the particular moment. This implies also the usage of the correct orbital parameters, e.g. inclination, true anomaly at start etc.

Nevertheless, to get an impression of the result magnitudes to be expected and to get the feeling of the usage, three independent orbits can be used. This is also the case for the following orbit descriptions.

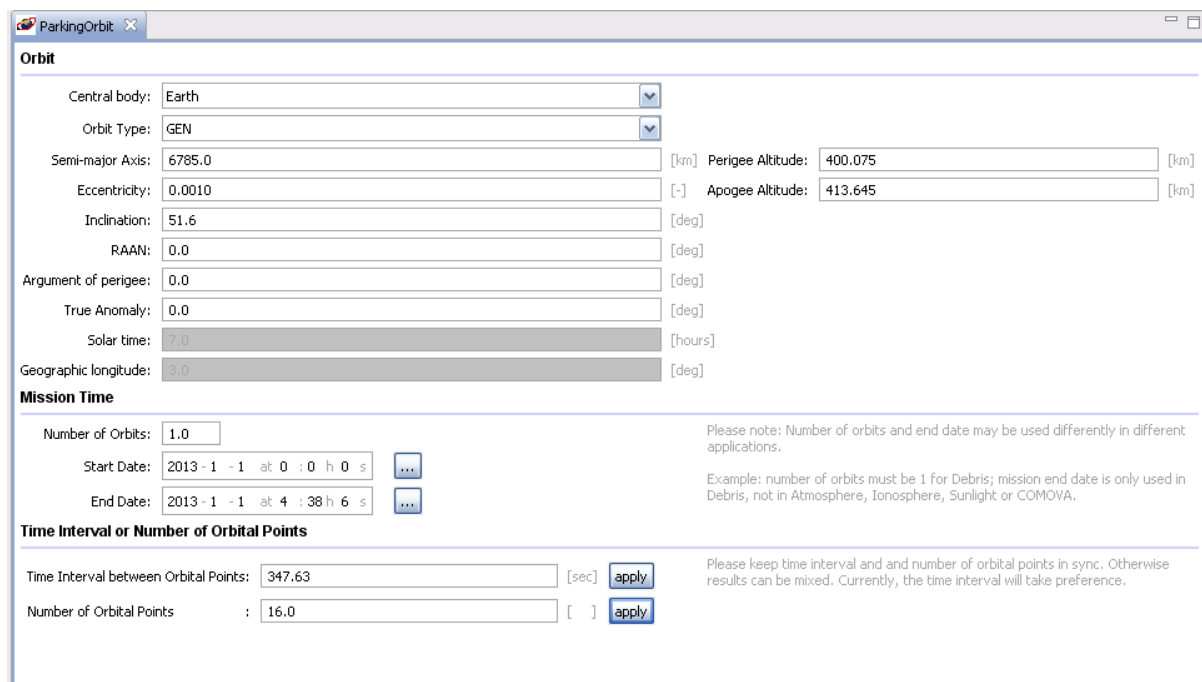
3.1.1.1 The Parking Orbit

Before the start of the journey to Moon the spacecraft is placed in an orbit around the Earth. A possibility is the low Earth orbit of the ISS. The basic values of such an orbit can be:

- Central body (centre of motion): Earth
- Semi-major axis = 6785 km
- Eccentricity = 0.001
- Inclination = 51.6 deg
- Other angles = 0.0 deg

The mission starts on January the 1, 2013 at 0:00. The orbital period of the orbit is ca. 92.7 minutes, let's say that 3 orbits are completed before starting the transfer, which means a mission duration of 278.1 minutes for the LEO. Thus the end epoch is January the 1, 2013 at 4:38 and 6 seconds.

The mission file can be created as described in /1/ section 3.1. The filename *ParkingOrbit* will be used in the text to identify the orbit. Figure 3.2 shows the mission editor with the parameters applied as introduced before.



The screenshot shows a software window titled "ParkingOrbit" with a tab "Orbit". The interface is divided into several sections:

- Orbit Parameters:**
 - Central body: Earth (dropdown)
 - Orbit Type: GEN (dropdown)
 - Semi-major Axis: 6785.0 [km]
 - Eccentricity: 0.0010 [-]
 - Inclination: 51.6 [deg]
 - RAAN: 0.0 [deg]
 - Argument of perigee: 0.0 [deg]
 - True Anomaly: 0.0 [deg]
 - Solar time: 7.0 [hours]
 - Geographic longitude: 3.0 [deg]
 - Perigee Altitude: 400.075 [km]
 - Apogee Altitude: 413.645 [km]
- Mission Time:**
 - Number of Orbits: 1.0
 - Start Date: 2013 - 1 - 1 at 0 : 0 h 0 s
 - End Date: 2013 - 1 - 1 at 4 : 38 h 6 s

Please note: Number of orbits and end date may be used differently in different applications.
Example: number of orbits must be 1 for Debris; mission end date is only used in Debris, not in Atmosphere, Ionosphere, Sunlight or COMOVA.
- Time Interval or Number of Orbital Points:**
 - Time Interval between Orbital Points: 347.63 [sec]
 - Number of Orbital Points: 16.0 []

Please keep time interval and number of orbital points in sync. Otherwise results can be mixed. Currently, the time interval will take preference.

Figure 3.2: Parking orbit parameters

Figure 3.1 shows that in a realistic scenario the first phase of a lunar mission is more complex than just one orbit. The mission phase is often composed of multiple orbits with different eccentricities and inclinations. Reasons for such an approach are the minimisation of the required propellant and also of the failure risk. This approach can be modelled by definition of new mission files, with according orbits and mission times, after each firing of the thrusters. The first phase of the lunar mission would be than described by multiple mission files, instead of one, the results of which have to be combined in the manual post-process.

3.1.1.2 Transfer Orbit

The transfer orbit is abstracted by a highly elliptic orbit around the Earth, with the perigee at the LEO orbit and the apogee at ca. the distance to the Moon.

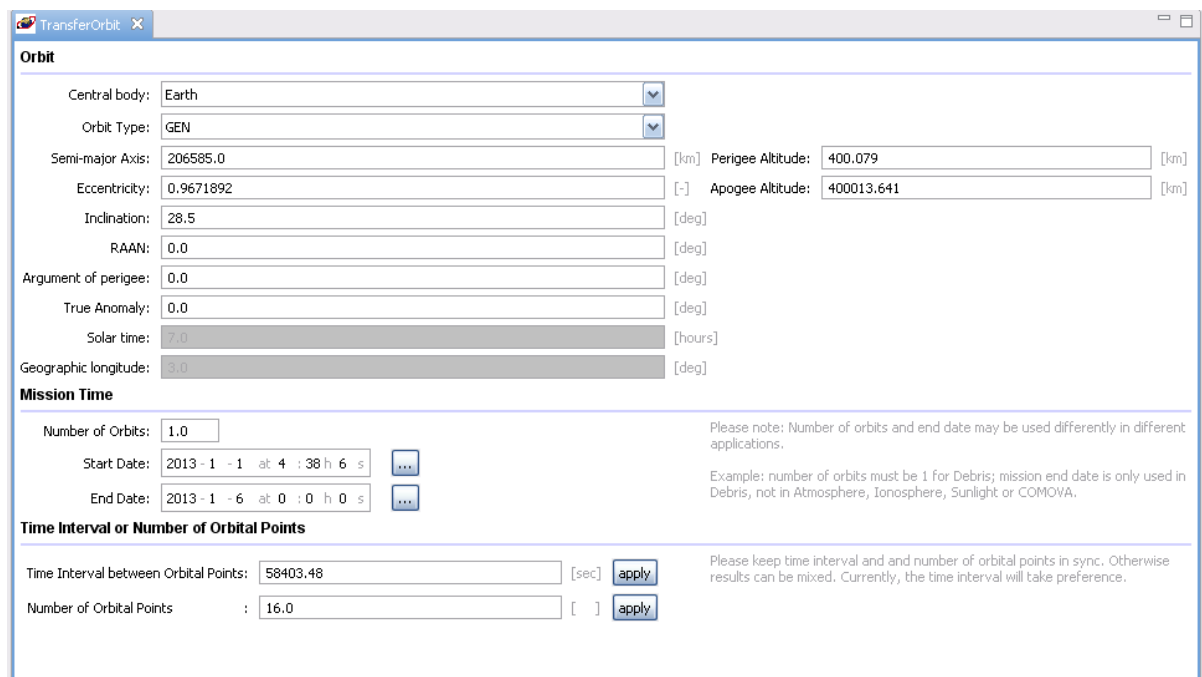
Possible parameters for a representative HEO are:

- Central body (centre of motion): Earth
- Semi-major axis = 206585 km
- Eccentricity = 0.9671892
- Inclination = 28.5 deg
- Other angles = 0.0 deg

This trajectory has nearly the same perigee as the one of the LEO.

The start time for the transfer orbit is the mission end epoch of the LEO mission file, i.e. January the 1, 2013 at 4:38 and 6 seconds. A HEO with the introduced parameters has a period of ca. 10.82 days. Thus the half of the period (only the way to Moon) would be for a Hohmann transfer ca. 5.41 days. Let's say the transfer duration is more than 5 days but shorter than the duration of a Hohmann transfer. The end epoch for the second phase of the lunar mission, the transfer is thus arbitrary adjusted to January the 6, 2013 at 0:00.

The mission file can be created as described in /1/ section 3.1. The filename *TransferOrbit* will be used in the text to identify the orbit. The Figure 3.3 show the mission editor with the parameters applied as introduced before.



The screenshot shows a software window titled "TransferOrbit" with a tab "TransferOrbit". The interface is divided into several sections:

- Orbit:**
 - Central body: Earth (dropdown)
 - Orbit Type: GEN (dropdown)
 - Semi-major Axis: 206585.0 [km]
 - Eccentricity: 0.9671892 [-]
 - Inclination: 28.5 [deg]
 - RAAN: 0.0 [deg]
 - Argument of perigee: 0.0 [deg]
 - True Anomaly: 0.0 [deg]
 - Solar time: 7.0 [hours]
 - Geographic longitude: 3.0 [deg]
 - Perigee Altitude: 400.079 [km]
 - Apogee Altitude: 400013.641 [km]
- Mission Time:**
 - Number of Orbits: 1.0
 - Start Date: 2013 - 1 - 1 at 4 : 38 h 6 s
 - End Date: 2013 - 1 - 6 at 0 : 0 h 0 s

Please note: Number of orbits and end date may be used differently in different applications.
Example: number of orbits must be 1 for Debris; mission end date is only used in Debris, not in Atmosphere, Ionosphere, Sunlight or COMOVA.
- Time Interval or Number of Orbital Points:**
 - Time Interval between Orbital Points: 58403.48 [sec]
 - Number of Orbital Points: 16.0 []

Please keep time interval and number of orbital points in sync. Otherwise results can be mixed. Currently, the time interval will take preference.

Figure 3.3: Transfer orbit parameters

The Figure 3.1 indicates that correction manoeuvres can be performed during the transfer. Such case can also be modelled by defining different mission files with according orbits and durations before and after the manoeuvre.

3.1.1.3 Target Orbit

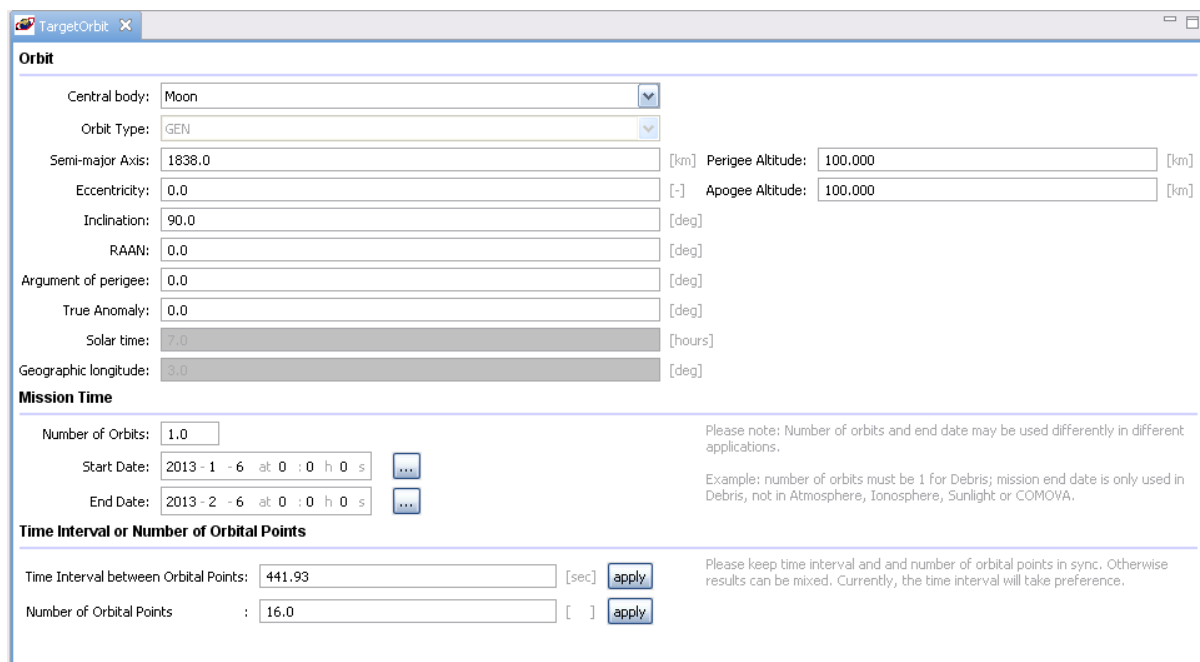
The target orbit is the desired trajectory around the Moon. Polar tracks with low altitudes are often aimed lunar orbits. Parameters for such orbit can be:

- Central body (centre of motion): Moon
- Semi-major axis = 1838 km
- Eccentricity = 0.0
- Inclination = 90 deg

- Other angles = 0.0 deg

The starting epoch of this phase of the lunar mission is the end epoch of the previous phase: January the 6, 2013 at 0:00. As already mentioned this is only an example, so let arbitrary say that the mission duration is one month, thus the end date is February the 6, 2013 at 0:00.

The mission file can be created as described in /1/ section 3.1. The filename *TargetOrbit* will be used in the text to identify the orbit. The Figure 3.4 show the mission editor with the parameters applied as introduced before.



The screenshot shows a software window titled "TargetOrbit" with the following parameters and controls:

- Orbit Section:**
 - Central body: Moon
 - Orbit Type: GEN
 - Semi-major Axis: 1838.0 [km]
 - Eccentricity: 0.0 [-]
 - Inclination: 90.0 [deg]
 - RAAN: 0.0 [deg]
 - Argument of perigee: 0.0 [deg]
 - True Anomaly: 0.0 [deg]
 - Solar time: 7.0 [hours]
 - Geographic longitude: 3.0 [deg]
 - Perigee Altitude: 100.000 [km]
 - Apogee Altitude: 100.000 [km]
- Mission Time Section:**
 - Number of Orbits: 1.0
 - Start Date: 2013 - 1 - 6 at 0 : 0 h 0 s
 - End Date: 2013 - 2 - 6 at 0 : 0 h 0 s
- Time Interval or Number of Orbital Points Section:**
 - Time Interval between Orbital Points: 441.93 [sec]
 - Number of Orbital Points: 16.0

Additional text in the interface includes: "Please note: Number of orbits and end date may be used differently in different applications." and "Example: number of orbits must be 1 for Debris; mission end date is only used in Debris, not in Atmosphere, Ionosphere, Sunlight or COMOVA." and "Please keep time interval and and number of orbital points in sync. Otherwise results can be mixed. Currently, the time interval will take preference."

Figure 3.4: Target orbit parameters

The Figure 3.1 depicts that in fact the targeted orbit is not reached immediately. After the capture of the spacecraft by the Moon the trajectory is corrected and reached via multiple thrusters impulses. Thus also the last phase of the mission can be modelled in more detail by defining various mission files, which describes the orbits between the firings.

3.1.2 Trajectory File

The trajectory file allows the user, in contrary to the abstraction of a lunar mission, to define the whole track of the lunar mission in one file using the maximum of 100 orbital points / state vectors. If more orbital points than the maximum are contained in the file, only the first 100 points can be used. In this case a warning is shown in the console view.

The Figure 3.5 provides an example of a trajectory file, which represents the half of a highly elliptical orbit around Earth by 9 orbital points. The apogee is at the Moon distance and the last two points are in the in the sphere of influence of the Moon, thus they are interpreted as orbital points of lunar orbits.

```

*****
* Example trajectory file.
*
* Head defines the coordinate system and the epoch format:
* Possible origin bodies: (currently)
*   Earth
*   Moon
* Origin is optional, if it is missing, Earth is assumed to be the origin body
*
* Possible target bodies: (currently)
*   Earth
*   Moon
*
* Possible coordinate system:
*   ECI: Earth Centered Inertial (EME2000)
*   LCI: Lunar Centered Inertial (Moon_J2000) Mean pole (z),
*       IAU node of J2000.0 (x) System
* Possible epoch format:
*   MJD: Modified Julian Day 1950
*   UTC: CCSDS ASCII time code &: YYYY-MM-DDThh:mm:ss.d->d
*
* Body provides the state vectors with appropriate epoch
* Each state vector represents an orbital point
* Format: epoch X Y Z X_dot Y_dot Z_dot
*****
Origin: Earth
Target: Moon
Coordinates: ECI
Epoch: MJD
*****
$$BEGIN
19798.48806713   -5167.031478526992   -5714.255328231473   0.164960105913E-12   4.624454537551   -8.794028563774   -1.656573672415
19799.12152037   148315.109921829629   -29170.226320218550   -22994.538303340792   1.759254471321   0.138532715746   -0.215028620544
19799.75497361   226226.195277236839   -18725.302590315150   -32003.944520032528   1.165682717708   0.221179444868   -0.127063680752
19800.38842685   280398.370256302762   -6027.124364110680   -37620.677676406325   0.896727320480   0.238308397627   -0.081729838999
19801.02188009   319467.564641026722   7060.320356665809   -41203.360972652750   0.601124790635   0.238235389148   -0.050731326032
19801.65533333   347003.923373181664   19916.583080674551   -43295.770668876990   0.410462982675   0.230658389569   -0.026541351840
19802.28878657   364833.864160855301   32226.446256702318   -44175.831010104965   0.244018276115   0.218532860558   -0.006080424525
19802.92223961   373947.690921552770   43771.471716275177   -43999.897190881798   0.090479017080   0.202768322262   0.012248515984
19803.55569306   374834.343681340863   54355.372132761142   -42855.701921501168   -0.057725707849   0.183352046203   0.029440347424
$$END

```

Figure 3.5: Trajectory file example

The trajectory file is composed of two parts (not considering the comments); the head and the body.

The head of the file provides information about the mission. The keywords for the kind of information are 'Origin', 'Target', 'Coordinates' and 'Epoch'. They are required to indicate which information is given. The colon is required as a marker for the start of the input. No predefined order for the provision of the information exist, it can theoretically be placed everywhere in the head (before \$\$BEGIN), but it is recommended to group the inputs for better readability. The input is handled not case sensitive.

Origin provides the information of the starting celestial body. Possible inputs are **Earth** and **Moon**. This input is optional; if no origin is given Earth is used as default.

Target provides the information of the journey's destination. Possible inputs are **Earth** and **Moon**. It is mandatory.

For trajectories around one celestial body (only Earth orbit or only lunar orbit), the origin and the target must be equal.

Coordinates provides the information of the coordinate frame that is used for the given state vectors. It is mandatory. Possible inputs are **ECI** and **LCI**. Thus the state vectors can be provided in Earth centred inertial (ECI) frame, which is EME2000, or in lunar centred inertial (LCI), sometimes also called Moon J2000. LCI is a mean pole (z-axis) IAU node (x-axis) at the epoch of J2000.0 system.

Epoch provides information of the time format used for the state vector epochs. It is mandatory. Possible inputs are **MJD** and **UTC**. MJD indicates that the epochs are given in

the modified Julian day 1950 format, thus a decimal value DDDDD.d→d. UTC implies the use of the CCSDS ASCII time code A, i.e. YYYY-MM-DDThh:mm:ss.d→d, for the definition of the epochs.

The body of the file provides the state vectors of the trajectory. It must be placed behind the head and has to start with the key '\$\$BEGIN'. If the key '\$\$END' is not found, a warning is directed to the console view, which states that the trajectory file could be incomplete, but the analysis is continued.

The listed state vectors have to be preceded by the according epoch in the predefined format (MJD or UTC) and follow the format: X Y Z X_dot Y_dot Z_dot. The state vector has to be in the predefined coordinate frame to get reasonable results.

Each state vector (with according epoch) is interpreted as an orbital point to be analysed. The maximum number of orbital points for the trajectory file is currently limited to 100.

The trajectory file can be modified with the ESABASE2 editor or every other text editor outside the application. To define a text file to be trajectory file and to be identified as such by ESABASE2 it must have the file type *.trajectory.

3.2 Debris Input Limitation for Lunar Mission

The use of the Debris solver and the generation of a debris file is described in the chapter 2. The intention of this section is to introduce the limitations associated with the analysis of lunar mission or the use of a trajectory file.

3.2.1 Specifics for Abstracted Lunar Mission

In the case of an abstracted lunar mission the individual mission steps, represented by the mission files, are analysed independently. Thus, theoretically, for the analysis of orbits in the vicinity of the Earth all model possibilities, except LunarMEM, are available. But if the results of the individual mission steps are to be combined (manual post-processing), they have to be coherent. This means coherent models should be used throughout the analysis of the mission. The use of the debris and meteoroid models can be seen as independently.

For each mission file a debris file can be defined due to the individual analysis. More practicable is the definition of one debris file for the analysis of orbits around the Earth and one debris file for the analysis of orbits around the Moon. The debris file around the Earth should define the debris (if desired) and meteoroid model to be used, e.g. MASTER 2009 and MEM. The debris file around Moon should define the meteoroid model to be used, e.g. LunarMEM.

3.2.1.1 Debris Models

Debris models are available only for the vicinity of the Earth and thus can be only applied to orbits around the Earth.

In the case that the debris environment should be considered only for the low Earth orbits than all debris models can be used. If the effects of the debris shall be considered for trajectories with semi-major axes about LEO (> 2000 km) also, than only the MASTER models are applicable. This practically excludes NASA90 (target altitude < 1000 km) and ORDEM2000

(target altitude < 2000 km) models, because using them will lead to neglecting of the debris environment in the vicinity of the Earth for the transfer orbits.

Three versions of the MASTER model are available. MASTER 2001 is the oldest version and based on outdated data, but it is fast and thus can be handy sometimes. MASTER 2005 is the successor of MASTER 2001 and based on real data up to 2005. It is applicable to the vicinity of the Earth up to 1000 km above the geosynchronous orbit. The limitation of MASTER 2005 is the provision of mission (averaged) results only. Due to the averaged results it should be ensured that the examined mission part is in the range of validity of MASTER 2005, otherwise not existing debris fluxes will be provided. Furthermore the debris analysis results are not tailored to the considered arc of the track. The best choice is the successor of MASTER 2005, the MASTER 2009 model. It is based on the most up-to-date data (real data up to 2009) of all five debris models. Furthermore it provides data for the each orbital point, besides the averaged mission data. Thus data for individual points of interests can be extracted from the results. If only a part of a transfer orbit is considered and analysed using MASTER 2009 the desired results shall be extracted from the orbital points included in the arc.

Summarising the properties of the debris models it can be concluded that MASTER 2009 is the preferable choice due to its actuality and the range of validity for the abstracted lunar mission. If the transfer orbit is represented by only one mission file as in *TransferOrbit*, then also MASTER 2005 can be used, which show a better performance. If the transfer trajectory is represented by multiple mission files, then MASTER 2009 shall be used and the results of individual orbital points included in the considered arc shall be combined to achieve the desired result for the arc.

3.2.1.2 Meteoroid Models

The ESABASE2/Debris application allows the use of four alternative meteoroid models; Grün, Divine-Staubach, MEM and LunarMEM. But the four can be reduced to three, because LunarMEM is a MEM version tailored to the vicinity of the Moon.

For the analysis of orbits around the Earth the models Grün, Divine-Staubach and MEM are available. For the analysis of lunar trajectories Grün and LunarMEM are available. To achieve coherent results for the whole track the models used for Earth and Moon vicinities shall be also coherent. Therefore either Grün model for both Earth and Moon part of the orbit or MEM for Earth and LunarMEM for Moon part of the orbit shall be used. Additionally to the missing equivalent for the lunar trajectories, the Divine-Staubach model application is limited to 1000 km above the geosynchronous orbit.

If the Grün model is applied for the analysis of a lunar mission it is recommended to use the Taylor HRMP velocity distribution. It provides the most reasonable results compared with other models. The Figure 3.6 shows the activation of the Taylor HRMP velocity distribution for the Grün model via the 'Edit' button.

The available NASA90 velocity distribution is designed for the vicinity of the Earth.

The consideration of the focusing effect of the celestial body for constant velocity disregards the gravity constant of the body. This leads to overestimation of the flux for lunar orbits.

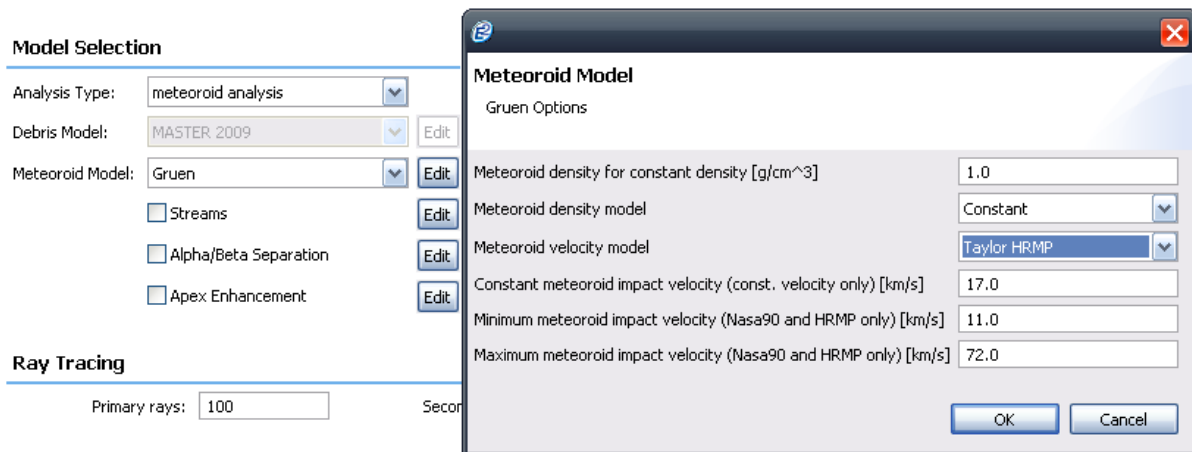


Figure 3.6: Taylor HRMP activation with the Edit button for Grün

3.2.2 Specifics for Trajectory File

The trajectory file can be combined with a limited number of models. The available models are selected according to the advantages and disadvantages described in 3.2.1.

Due to the possibility to tailor the results to desired arcs and handling a higher variety of orbits, the MASTER 2009 model is available for the debris analysis.

The models Grün, MEM and LunarMEM can be combined with a trajectory file. Thereby the choice of MEM or of LunarMEM means the same to the software. It automatically switches from MEM to LunarMEM if the centre of motion is identified to be the Moon.

Since the debris and the mission files are independent, a try for each pair of files can be given. Non-compatible files will result in an error message, which should describe the problem.

3.3 Perform an Analysis

Three mission files were presented in section 3.1 abstracting the lunar mission and two debris files were suggested (Earth: MASTER 2009, MEM and Moon: LunarMEM) in section 3.2. Also a geometry is required for a geometry analysis. A sample geometry is generated in ESABASE2/Framework software user manual /1/ in section 3.2, this can be used for the analysis.

How to perform an analysis is described in section 2.3.1. For the case of an abstracted lunar mission three analyses have to be performed. Each with the same geometry but for the two analyses with *ParkinOrbit* and *TransferOrbit* mission files the debris file for orbits around the Earth has to be used. The last analysis is combined of the geometry the *TargetOrbit* and the debris files for the orbits around the Moon.

After the performance of all three analysis the results need to be combined in a manual post-processing to get the results of the whole mission.

To use a trajectory file it can be simply chosen in the drop down menu of the mission files as shown in Figure 3.7.

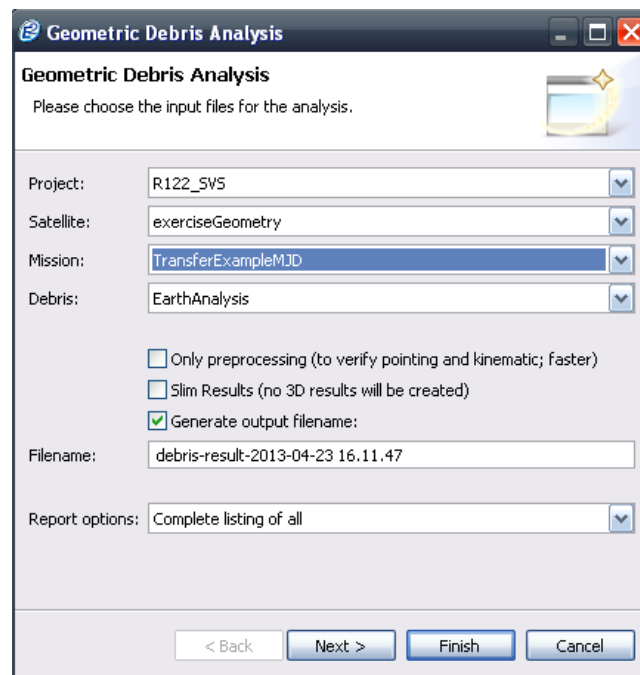


Figure 3.7: Choosing a trajectory file